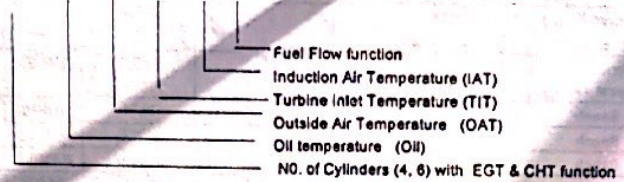
ELIGIBLE PART NUMBERS for EDM-760 PN 760000

Note (2) suffix "•" denotes turbocharged engines.

Model designation system by part number for EDM-760

760000-[ ]C -O[ ] -A[ ] -T[ ] -I[ ] -[ ]F



Example: 760000 - (6COAI) is indicated as such on the TSO label

760000 - 6C - 0 [•] -A [•] -T [ ] -I [•] - F [•]

EDM 760 Model

- 6 Cylinder
- Oil Temperature function
- OAT, Outside Air function
- IAT, Induction Air function
- Fuel Flow

ELIGIBLE PART NUMBERS for 760000

Part number Series

4 4 Cylinder	6 6 Cylinder
760000-( )	760000-( )
4CF	6CF
4COF	6COF
4COAF	6COAF
4COATF	6COATF
4COTF	6COTF
4COTIF	6COTIF
4COAIF	6COAIF
4CATIF	6CATIF
4CAF	6CAF
4CATF	6CATF
4CAIF	6CAIF
4CTIF	6CTIF
4CTF	6CTF
4CIF	6CIF
4COATIF	6COATIF

FAA APROVED:

Acting Manager, Special Certification Branch
Seattle Aircraft Certification Office

Amendment Date: June 11, 2001

United States of America

Department of Transportation—Federal Aviation Administration

Supplemental Type Certificate

Number SA00729SE

This certificate, issued to J. P. Instruments
P.O. Box 7033
Huntington Beach, CA 92646

*certifies that the change in the type design for the following product with the limitations and conditions thereto as specified hereon meets the airworthiness requirements of Part * of the * Regulations.*

Original Product—Type Certificate Number: *See attached FAA Approved Model List (AML)
Make: No. SA00729SE for a list of approved airplane
Model: models and applicable airworthiness regulations.

Description of the Type Design Change: Installation of twin temperature indicating system with fuel flow in accordance with J.P. Instruments (JPI) Installation manual Report No. 760, Revision -, dated 7/20/99, or later FAA approved revision.

Limitations and Conditions: Approval of this change in type design applies to the aircraft models listed on the AML only. This approval should not be extended to other aircraft of these models on which other previously approved modifications are incorporated unless it is determined that the relationship between this change and any of those other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that aircraft. A copy of this certificate, Airplane Flight Manual Supplement No. 760-1 dated August 31, 1999, or later FAA approved revision, and FAA Approved Model List (AML) No. SA00729SE must be maintained as part of the permanent records for the modified aircraft.

Cylinder head, oil, turbine inlet and/or exhaust gas temperature, fuel flow equipment, tachometer instruments, and manifold pressure instruments required by the original type design, or if required by other FAA approval, must remain installed.

If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: June 18, 1999
Date of issuance: August 31, 1999

Date reissued:
Date amended:

By direction of the Administrator

(Signature)

Acting Manager, Seattle Aircraft Certification Office
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding $1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

FAA Form 8110-2(10-88)

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

REMOVED EQUIPMENT:
AUTOMATIC PRESSURE ALTITUDE DIGITIZER

INSTALLED EQUIPMENT:
INSTALLED A GARMIN GDL69A XM WEATHER AND RADIO RECEIVER I/A/W MANUFACTURERS INSTALLATION MANUAL P/N 190-00355-02 REVISION D DATED 9/05, GARMIN MASTER DATA LIST, P/N 005-C0217-00, REV C, DATED 8/05 AND STC # SA01487SE PER THE AML AND AC 43.13-1B CHAPTER 12 SECTION 2 PARAGRAPH 12-8. THE STC WAS USED WITH WRITTEN PERMISSION FROM GARMIN ACCORDING TO THE GARMIN GDL69/69A INSTALLATION MANUAL.

INSTALLED A GARMIN GA 55 XM SATELLITE RADIO ANTENNA I/A/W MANUFACTURERS INSTALLATION MANUAL P/N 190-00355-02 REVISION D, DATED 9/05 AND STC # SA01487SE PER THE AML AND AC43.13-2A CHAPTER 3 ANTENNA INSTALLATIONS PARAGRAPH 36 AND PARAGRAPH 44.

INSTALLED A GARMIN GMX 200 STANDARD MULTI-FUNCTION DISPALY I/A/W MANUFACTURERS INSTALLATION MANUAL P/N 190-00607-04 REVISION B DATED 7/10/06, GARMIN STC MASTER DATA LIST, P/N 005-C0315-00, REVISION A, DATED 7/25/06 AND STC # SA01692SE PER THE AML. THE STC WAS USED WITH WRITTEN PERMISSION FROM GARMIN ACCORDING TO APPENDIX A OF THE GARMIN GMX 200 INSTALLATION MANUAL. THE GMX 200 IS INTERFACED VIA RS-232 SERIAL DATA TO THE GNS 530 GPS TO RECEIVE POSITION INFORMATION. THE MFD DISPLAY IS FOR SUPPLEMENTAL (VFR) NAVIGATION ONLY AND IS NOT TO BE UTILIZED FOR IFR NAVIGATION. A FAA APPROVED FLIGHT MANUAL SUPPLEMENT WAS PLACED INSIDE THE AIRCRAFT.

ICA'S ARE ATTACHED.

THIS INSTALLATION ALSO COMPLIES WITH AC43.13-1B CHAPTER 10 WEIGHT AND BALANCE, AC43.13-1B CHAPTER 11 AIRCRAFT ELECTRICAL SYSTEMS AND AC43.13-1B CHAPTER 12 AIRCRAFT AVIONICS SYSTEMS SECTION 2 PARAGRAPH 12-9.

********** NOTHING FOLLOWS **********

☑ Additional Sheets Are Attatched

*U.S.GPO:1994-568-012/00018

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished
(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N421MY	09/11/2012
Nationality and Registration Mark	Date

REMOVED THE FOLLOWING EQUIPMENT:
MANUFACTURE/MODEL
GARMIN GNS 530W GPS WAAS NAVIGATION AND GA 35 GPS WAAS ANTENNA

INSTALLED THE FOLLOWING EQUIPMENT;
MANUFACTURE/MODEL
GARMIN GWX 68 RADAR SYSTEM – 12"
GARMIN GTN 750 GPS WAAS NAVIGATION SYSTEM WITH A GA 35 GPS WAAS ANTENNA

INSTALLATION DESCRIPTION:

1. INSTALLED A GARMIN GTN 750 GPS/SBAS NAVIGATION SYSTEM WITH GARMIN GA 35 GPS WAAS ANTENNA. THE GA 35 GPS WAAS ANTENNA WAS INSTALLED IN THE SAME LOCATION AS THE REMOVED GA 35 GPS WAAS ANTENNA. THE STC WAS USED WITH WRITTEN PERMISSION FROM GARMIN ACCORDING TO THE STC PERMISSION LETTER TO USE STC SA02019SE-D FOR GARMIN MODEL GTN 6XX/7XX NAVIGATION SYSTEM. THE GTN 750 GPS/SBAS NAVIGATION SYSTEM DOES NOT INTERFERE WITH THE NORMAL OPERATION OF OTHER EQUIPMENT INSTALLED IN THE AIRCRAFT AS CONFIRMED BY GROUND AND FLIGHT TESTING. THE GTN 750 GPS/SBAS NAVIGATION SYSTEM IS PROTECTED WITH THE REQUIRED CIRCUIT BREAKERS MOUNTED IN THE AVIONICS BUSS. AN ELECTRICAL LOAD ANALYSIS HAS BEEN CALCULATED AND DETERMINED THAT THE TOTAL ELECTRICAL LOAD DOES NOT EXCEED 85A. THE GTN 750 GPS/SBAS NAVIGATION SYSTEM IS INTERFACED TO THE GARMIN GWX 68 RADAR SYSTEM, THE GARMIN GTX 330 MODE S TRANSPONDER, THE GARMIN GDL 69A XM WEATHER AND RADIO RECEIVER, THE GARMIN GMA 340 AUDIO PANEL, THE SANDEL SN3500 EHSI, THE KING KN 64 DME, THE JP INSTRUMENTS EDM 760 ENGINE MONITOR, THE SHADIN ADC AND TO THE S-TEC SYSTEM 55X AUTOPILOT THROUGHT THE SANDEL SN3500 EHSI.

2. INSTALLED A GARMIN GWX 68 WEATHER RADAR SYSTEM WITH A 12" PLATE. THE GARMIN GWX 68 IS INTERFACED TO THE GARMIN GTN 750 GPS/SBAS NAVIGATION SYSTEM FOR DISPLAY AND CONTROL.

REFERENCES:

1. GARMIN GTN 6XX/7XX AML STC INSTALLATION MANUAL P/N 190-01007-A3 REVISION 3 DATED 5/23/11
GARMIN STC# SA02019SE-D PER THE AML

2. GARMIN INSTALLATION MANUAL P/N 190-00286-01 REVISION E DATED 6/1/10
AC43.13-2B CHAPTER 2 COMMUNICATION, NAVIGATION AND EMERGENCY LOCATOR TRANSMITTER SYSTEM INSTALLATIONS PAGE 10 – PARAGRAPH 202, PAGE 18 – PARAGRAPH 207, PAGE 19 – PARAGRAPH 208 AND 209, AND PAGE 21 – PARAGRAPH 210
THE GARMIN GWX 68 HAS BEEN STC'D IN A RAYTHEON AIRCRAFT COMPANY 65-B80, A65, 70, 65-90, 65-A90, B90, C90, C90A, C90GT AND E90 UNDER STC SA01670SE.

NOTES:

A. THIS INSTALLATION HAS BEEN FUNCTIONALLY CHECKED AND FOUND TO OPERATE NORMALLY AND IS IN COMPLIANCE WITH FAR 23.1301, FAR 23.1309 AND CHECKED I/A/W FAR 23.1431 FOR OPERATING SATISFACTORILY AND HAD NO ADVERSE AFFECTS NOTED ON ANY OTHER COMPONENTS INSTALLED IN THE AIRCRAFT.

B. THIS ITEM WILL BE MAINTAINED I/A/W THE MANUFACTURES MAINTENANCE INSTRUCTIONS AND INSPECTED I/A/W FAR 43 APPENDIX D(j).

C. INSTRUCTIONS FOR CONTINUED AIRWORTHINESS ARE ATTACHED

D. AN FAA APPROVED AIRPLANE FLIGHT MANUAL SUPPLEMENT FOR THE GTN 750 GPS/SBAS NAVIGATION SYSTEM P/N 190-01007-A2 REVISION 1 DATED 3/18/2011 WAS PLACED INSIDE THE AIRCRAFT.

☒ Additional Sheets Are Attached

Electronic Form (PDF)

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished
(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N421MY	09/11/2012
Nationality and Registration Mark	Date

E. AIRCRAFT WEIGHT AND BALANCE COMPLETED AND EQUIPMENT LIST HAS BEEN REVISED TO REFLECT THE MODIFICATIONS AND HAVE BEEN INSERTED IN THE AIRCRAFT POH/AFM - AC43.13-1B CHAPTER 10 WEIGHT AND BALANCE PAGE 10-14 PARAGRAPH 10-20.

F. TO INSURE INTEGRITY, THIS SYSTEM MUST BE CHECKED I/A/W THE MANUFACTURER'S POST MAINTENANCE PROCEDURES FOLLOWING ANY MANITENANCE PERFORMED ON THE SYSTEM.

G. THIS INSTALLATION ALSO COMPLIES WITH:
AC43.13-1B CHAPTER 11 AIRCRAFT ELECTRICAL SYSTEMS
SECTION 1 – INSPECTION AND CARE OF ELECTRICAL SYSTEMS
SECTION 3 – INSPECTION OF EQUIPMENT INSTALLATION
SECTION 4 – INSPECTION OF CIRCUIT-PROTECTION DEVICES
SECTION 5 – ELECTRICAL WIRE RATING
SECTION 6 – AIRCRAFT ELECTRICAL WIRE SELECTION
SECTION 7 – TABLE OF ACCEPTABLE WIRES
SECTION 8 – WIRING INSTALLATION INSPECTION REQUIREMENTS

AC43.13-1B CHAPTER 12 AIRCRAFT AVIONICS SYSTEMS
SECTION 2 PARAGRAPH 12-9A – INSPECTION OF AVIONICS SYSTEMS
SECTION 2 PARAGRAPH 12-9B – INSPECT ANTENNAS FOR:

********** NOTHING FOLLOWS **********

☐ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

REMOVED EQUIPMENT:

CESSNA FUEL FLOW, CESSNA ALTIMETER, CESSNA TURN COORDINATOR, ALCOR AVIATION INC. EGT, ARC S-530A MODE SELECTOR, ARC DT-520A TRIM SENSOR, ARC PA-520A ACTUATOR, ARC C-830FD CONTROL UNIT, ARC PA-500A ACTUATOR, ARC PA-500A ACTUATOR, ARC CA-530FD COMPUTER AMPLIFIER, ARC FLIGHT DIRECTOR HORIZON, AND A FAN

INSTALLED EQUIPMENT:

INSTALLED AN S-TEC SYSTEM 55X AUTOPILOT WITH AUTOTRIM I/A/W S-TEC INSTALLATION MODEL ST-559 BULLETIN NO. 659 REVISION 6 DATED 9-16-02, S-TEC INSTALLATION MODEL TK-559 BULLETIN NO. 659-1 REVISION 6 DATED 9-16-02 AND S-TEC GENERAL INSTALLATION INFORMATION BULLETIN NO. 600 REVISION 20 DATED 11-12-03. INSTALLED AN S-TEC ST-1000 ALTITUDE SELECTOR / ALERTER I/A/W S-TEC ST-1000 ALTITUDE SELECTOR / ALERTER BULLETIN NO. 1000 REVISION 2 DATED 1-23-06. INSTALLED AN S-TEC SYSTEM 55X AUTOPILOT ANNUNCIATOR I/A/W S-TEC ANNUNCIATOR SYSTEM 55/55X AUTOPILOT ST-645 BULLETIN NO. 645 REVISION 7 DATED 2-7-01. INSTALLED AN S-TEC SINGLE CUE FLIGHT DIRECTOR/STEERING HORIZON I/A/W S-TEC SINGLE CUE FLIGHT DIRECTOR/STEERING HORIZON ST-361 BULLETIN NO. 361 REVISION 15 DATED 2-13-01. ALL OF THE ABOVE ITEMS WERE INSTALLED I/A/W S-TEC STC# SA8890SW-D.

A FAA APPROVED FLIGHT MANUAL SUPPLEMENT WAS PLACED INSIDE THE AIRCRAFT FOR THE S-TEC SYSTEM 55X TWO AXIS AUTOMATIC FLIGHT GUIDANCE SYSTEM AND ONE FOR THE S-TEC SA-200 ALTITUDE SELECTOR/ALERTER.

ICA'S ARE ATTACHED.

THIS INSTALLATION ALSO COMPLIES WITH AC43.13-1B CHAPTER 10 WEIGHT AND BALANCE, AC43.13-1B CHAPTER 11 AIRCRAFT ELECTRICAL SYSTEMS AND AC43.13-1B CHAPTER 12 AIRCRAFT AVIONICS SYSTEMS SECTION 2 PARAGRAPH 12-9.

********** NOTHING FOLLOWS **********

☑ Additional Sheets Are Attatched

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

A. Installed J.P. Instruments EDM 760 with Fuel Flow Engine Monitor System. The existing fuel flow gage was removed and the EDM 760 installed in the same location. The EDM 760 engine monitor modification to the aircraft was performed in reference to STC Number SA00729SE.

B. A field approval is requested for a deviation from the STC of removing the existing fuel flow indicator in order remove fuel carrying lines from the cockpit. Cessna Aircraft started removing fuel lines from the cockpit and installed electronic fuel flow indicating systems in 1978 on 400 series aircraft.

C. All components were installed, wired and secured in accordance with instructions provided within J.P. Instruments Installation Manual Report #760 and appendix A revised 7/20/99.

D. The aircraft weight and balance, and the aircraft equipment list were amended in accordance with AC 43.13-1B Chapter 10.

E. All inspection records and other documents pertaining to this major alteration are on file at repair station under Work Order 16749.

F. ICA is attached.

********** NOTHING FOLLOWS **********

☑ Additional Sheets Are Attatched

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N421MY	3/28/2007
Nationality and Registration Mark	Date

Validated that the previous installations of one GNS 430 and one GNS 530 was installed I/A/W Garmin instructions. Verified this aircraft and all interfaced equipment are covered under the STC AML. Both units were removed and upgraded to GNS 430W and GNS 530W units. The existing locations of the two units were determined to meet the field-of-view requirements without the need for external annunciation. The existing RG 58 GPS antenna cable was removed and replaced with RG 142 cable. The existing wiring and shielding was inspected and determined to be I/A/W the STC AML installation data. The two existing GA 56 antennas were removed and replaced with two GA 35 antennas using the approved mounting previsions of the previous installation.

A summary of the modification done to the aircraft is as follows:

1. Removed two (2) Garmin GA 56 Antennas, P/N 011-00134-00 and installed two (2) new GA 35 GPS/WAAS Antennas P/N 013-00235-00 using the provisions left behind from the standard antennas I/A/W with the Garmin STC Upgrade Installation Manual P/N 190-00357-06 Revision B and STC No. SA01933LA.
2. Removed Garmin GNS 430 P/N 011-00280-10 unit and installed Garmin GNS 430W P/N 011-01060-40 using the provision left behind from the standard GNS 430 unit. Installation done I/A/W with Garmin STC Upgrade Installation Manual P/N 190-00357-06 Revision B and STC No. SA01933LA.
3. The GNS 430W was configured identical to the original GNS 430 unit. Each interface was checked out I/A/W the 400W Series Installation Manual P/N 190-00356-02 Section 5. A copy of the checkout log was completed and included with the aircraft maintenance records.
4. Removed Garmin GNS 530 P/N 011-00550-00 unit and installed Garmin GNS 530W P/N 011-01064-45 using the provision left behind from the standard GNS 530 unit. Installation done I/A/W Garmin STC Upgrade Installation Manual P/N 190-00357-06 Revision B and STC No. SA01933LA.
5. The GNS 530W was configured identical to the original GNS 530 unit. Each interface was checked out I/A/W the 500W Series Installation Manual P/N 190-00357-02 Section 5. A copy of the checkout log was completed and included with the aircraft maintenance records.
6. Removed the Aircraft Flight Manual Supplement for the GNS 430 and installed a Garmin 400W Series AFMS P/N 190-00356-63, Revision B with FAA Approved date 12/21/2006 into the Aircraft Flight Manual.
7. Removed the Aircraft Flight Manual Supplement for the GNS 530 and installed a Garmin 500W Series AFMS P/N 190-00357-63, Revision B with FAA Approved date 12/21/2006 into the Aircraft Flight Manual.
8. Updated the aircraft Equipment List and Weight and Balance to reflect the new WAAS units. The current electrical load analysis remains valid since the new units draw the same or less current than the original units.

Instructions for Continued Airworthiness (ICA)

GNS 430W - Included 400W Series Instructions for Continued Airworthiness in the aircraft maintenance records.

GNS 530W - Included 500W Series Instructions for Continued Airworthiness in the aircraft maintenance records.

Note: These supersede ICAW data for the previously installed GNS 430 and GNS 530.

********** NOTHING FOLLOWS **********

[X] Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N421MY	4/4/2007
Nationality and Registration Mark	Date

A. Installed the following equipment and components:
   1. Sandel Avionics, Inc., SN3500 Navigation Display, P/N SN3500-002.
   2. Sandel Avionics, Inc., Mounting Clamp, P/N 61062.

B. The Sandel Avionics SN3500 is interfaced to the following equipment:
   1. Garmin International, GNS 530, COMM/NAV/GPS (Approved for Enroute, Terminal, and Precision Approach).
   2. Garmin International, GNS 430, COMM/NAV/GPS (Approved for Enroute, Terminal, and Precision Approach).
   3. Garmin International, GMA 340, Audio Panel for Marker Beacon.
   4. Cessna, 800 ADF System.
   5. L3 Avionics WX-500 Stormscope.
   6. S-TEC System 55X Autopilot.
   7. ARC RMI Indicator.

C. The SN3500 receives and processes GPS and VOR/LOC/GS navigation information for digital and waypoint display from the GNS 530 and GNS 430.

D. The SN3500 receives and processes marker beacon receiver information for illumination from the GMA 340 Audio Panel.

E. The SN3500 receives lightning information from the WX-500 Stormscope.

F. The SN3500 receives and processes magnetic heading for digital and graphic display from the ARC RMI.

G. Interference and functional tests and inspections were accomplished with reference to Advisory Circular 23.1311-1A.

H. A system design and analysis was conducted with reference to Advisory Circular 23.1309-1.

I. Federal Aviation Regulations, 23.1301, 23.1309(a), (b) and (d), 23.1311, 23.1321(a), (b) and (d), 23.1322, 23.1327(a), 23.1331, 23.1351, 23.1357(a)-(d), 23.1365, 23.1381, 23.1529, and 23.1581 were the basis for compliance.

J. Installation approval is sought with reference to Flight Standards Information Bulletin, FSAW 95-09(D) (Amended), titled "Electronic Horizontal Situation Indicator (EHSI) Approvals".

K. The aircraft equipment list, and weight and balance were revised and recorded within the aircraft maintenance records.

L. All pertinent records of this alteration are on file at Howard Aviation, Inc. Repair Station TO3R8874L under Work Order # 17740.

ICA is attached.

This installation also complies with AC43.13-1B Chapter 11 Aircraft Electrical Systems and AC43.13-1b Chapter 12 Aircraft Avionics Systems Section 2 Paragraph 12-9.

********** NOTHING FOLLOWS **********

☒ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished
(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N421MY	12/19/2008
Nationality and Registration Mark	Date

REMOVED EQUIPMENT:
N/A

INSTALLED EQUIPMENT:
INSTALLED A SHADIN DIGIDATA AIRDATA COMPUTER AT FS 8 IN THE NOSE AVIONICS BAY I/A/W MANUFACTURERS INSTALLATION MANUAL IM2803 REVISION K DATED 2/25/05, AC43.13-1B CHAPTER 12 SECTION 4 PARAGRAPH 52, AC43.13-2B CHAPTER 2, PAGE 12 PARAGRAPH 204, PAGE 14 PARAGRAPH 205, PAGE 18 PARAGRAPH 207 A-D AND F AND CHAPTER 11 PAGE 125 PARAGRAPH 1104 B & D. THE DIGIDATA AIRDATA COMPUTER IS INTERFACED TO THE EXISTING HEADING SYSTEM AND THE GARMIN GNS 430W AND 530W TO PROVIDE AIRDATA.

INSTRUCTIONS FOR CONTINUED AIRWORTHINESS ARE ATTACHED.

THIS INSTALLATION ALSO COMPLIES WITH AC43.13-1B CHAPTER 10 WEIGHT AND BALANCE PAGE 10-14 PARAGRAPH 10-20.

*** NOTHING FOLLOWS ***

☐ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished
(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N421MY	09/11/2012
Nationality and Registration Mark	Date

REMOVED THE FOLLOWING EQUIPMENT:
MANUFACTURE/MODEL
GARMIN GNS 530W GPS WAAS NAVIGATION AND GA 35 GPS WAAS ANTENNA

INSTALLED THE FOLLOWING EQUIPMENT:
MANUFACTURE/MODEL
GARMIN GWX 68 RADAR SYSTEM – 12"
GARMIN GTN 750 GPS WAAS NAVIGATION SYSTEM WITH A GA 35 GPS WAAS ANTENNA

INSTALLATION DESCRIPTION:

1. INSTALLED A GARMIN GTN 750 GPS/SBAS NAVIGATION SYSTEM WITH GARMIN GA 35 GPS WAAS ANTENNA. THE GA 35 GPS WAAS ANTENNA WAS INSTALLED IN THE SAME LOCATION AS THE REMOVED GA 35 GPS WAAS ANTENNA. THE STC WAS USED WITH WRITTEN PERMISSION FROM GARMIN ACCORDING TO THE STC PERMISSION LETTER TO USE STC SA02019SE-D FOR GARMIN MODEL GTN 6XX/7XX NAVIGATION SYSTEM. THE GTN 750 GPS/SBAS NAVIGATION SYSTEM DOES NOT INTERFERE WITH THE NORMAL OPERATION OF OTHER EQUIPMENT INSTALLED IN THE AIRCRAFT AS CONFIRMED BY GROUND AND FLIGHT TESTING. THE GTN 750 GPS/SBAS NAVIGATION SYSTEM IS PROTECTED WITH THE REQUIRED CIRCUIT BREAKERS MOUNTED IN THE AVIONICS BUSS. AN ELECTRICAL LOAD ANALYSIS HAS BEEN CALCULATED AND DETERMINED THAT THE TOTAL ELECTRICAL LOAD DOES NOT EXCEED 85A. THE GTN 750 GPS/SBAS NAVIGATION SYSTEM IS INTERFACED TO THE GARMIN GWX 68 RADAR SYSTEM, THE GARMIN GTX 330 MODE S TRANSPONDER, THE GARMIN GDL 69A XM WEATHER AND RADIO RECEIVER, THE GARMIN GMA 340 AUDIO PANEL, THE SANDEL SN3500 EHSI, THE KING KN 64 DME, THE JP INSTRUMENTS EDM 760 ENGINE MONITOR, THE SHADIN ADC AND TO THE S-TEC SYSTEM 55X AUTOPILOT THROUGHT THE SANDEL SN3500 EHSI.

2. INSTALLED A GARMIN GWX 68 WEATHER RADAR SYSTEM WITH A 12" PLATE. THE GARMIN GWX 68 IS INTERFACED TO THE GARMIN GTN 750 GPS/SBAS NAVIGATION SYSTEM FOR DISPLAY AND CONTROL.

REFERENCES:

1. GARMIN GTN 6XX/7XX AML STC INSTALLATION MANUAL P/N 190-01007-A3 REVISION 3 DATED 5/23/11
GARMIN STC# SA02019SE-D PER THE AML

2. GARMIN INSTALLATION MANUAL P/N 190-00286-01 REVISION E DATED 6/1/10
AC43.13-2B CHAPTER 2 COMMUNICATION, NAVIGATION AND EMERGENCY LOCATOR TRANSMITTER SYSTEM INSTALLATIONS PAGE 10 – PARAGRAPH 202, PAGE 18 – PARAGRAPH 207, PAGE 19 – PARAGRAPH 208 AND 209, AND PAGE 21 – PARAGRAPH 210
THE GARMIN GWX 68 HAS BEEN STC'D IN A RAYTHEON AIRCRAFT COMPANY 65-B80, A65, 70, 65-90, 65-A90, B90, C90, C90A, C90GT AND E90 UNDER STC SA01670SE.

NOTES:

A. THIS INSTALLATION HAS BEEN FUNCTIONALLY CHECKED AND FOUND TO OPERATE NORMALLY AND IS IN COMPLIANCE WITH FAR 23.1301, FAR 23.1309 AND CHECKED I/A/W FAR 23.1431 FOR OPERATING SATISFACTORILY AND HAD NO ADVERSE AFFECTS NOTED ON ANY OTHER COMPONENTS INSTALLED IN THE AIRCRAFT.

B. THIS ITEM WILL BE MAINTAINED I/A/W THE MANUFACTURES MAINTENANCE INSTRUCTIONS AND INSPECTED I/A/W FAR 43 APPENDIX D(j).

C. INSTRUCTIONS FOR CONTINUED AIRWORTHINESS ARE ATTACHED.

D. AN FAA APPROVED AIRPLANE FLIGHT MANUAL SUPPLEMENT FOR THE GTN 750 GPS/SBAS NAVIGATION SYSTEM P/N 190-01007-A2 REVISION 1 DATED 3/18/2011 WAS PLACED INSIDE THE AIRCRAFT.

☒ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished
(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N421MY	09/11/2012
Nationality and Registration Mark	Date

E. AIRCRAFT WEIGHT AND BALANCE COMPLETED AND EQUIPMENT LIST HAS BEEN REVISED TO REFLECT THE MODIFICATIONS AND HAVE BEEN INSERTED IN THE AIRCRAFT POH/AFM - AC43.13-1B CHAPTER 10 WEIGHT AND BALANCE PAGE 10-14 PARAGRAPH 10-20.

F. TO INSURE INTEGRITY, THIS SYSTEM MUST BE CHECKED I/A/W THE MANUFACTURER'S POST MAINTENANCE PROCEDURES FOLLOWING ANY MANITENANCE PERFORMED ON THE SYSTEM.

G. THIS INSTALLATION ALSO COMPLIES WITH:
AC43.13-1B CHAPTER 11 AIRCRAFT ELECTRICAL SYSTEMS
SECTION 1 – INSPECTION AND CARE OF ELECTRICAL SYSTEMS
SECTION 3 – INSPECTION OF EQUIPMENT INSTALLATION
SECTION 4 – INSPECTION OF CIRCUIT-PROTECTION DEVICES
SECTION 5 – ELECTRICAL WIRE RATING
SECTION 6 – AIRCRAFT ELECTRICAL WIRE SELECTION
SECTION 7 – TABLE OF ACCEPTABLE WIRES
SECTION 8 – WIRING INSTALLATION INSPECTION REQUIREMENTS

AC43.13-1B CHAPTER 12 AIRCRAFT AVIONICS SYSTEMS
SECTION 2 PARAGRAPH 12-9A – INSPECTION OF AVIONICS SYSTEMS
SECTION 2 PARAGRAPH 12-9B – INSPECT ANTENNAS FOR:

********** NOTHING FOLLOWS **********

☐ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished
(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Dated 04-24-06 Engine Previous Total: 3586.2

Engine crankcase modified per Dwg. 1514, Rev. R dated 6/29/05 I/A/W STC SE8338SW.

Relocated Turbo Oil Supply Line I/A/W RAM Dwg. No. 1224 Rev. H dated 11/18/03 and installed locknuts on cylinder attachment studs I/A/W Dwg. 1517, Rev. F dated 3/9/05 per STC SE8338SW.

TCM GTSIO-520 series engine crankshaft counterweights p/n 652833 are repaired in accordance with FAA approved procedures per RAM Drawing 2618, Rev. B dated 11/15/05. See engine log entry on Instructions for Continued Airworthiness.

Installation mechanic must complete Block 1 and 2 on reverse side and mail one copy to their local FSDO.

Negligible weight and balance change.

Customer furnished with FAA approved Overhaul and Parts Manual Supplements with Instructions for Continued Airworthiness for all alterations.

Pertinent details of the above installations are on file under Project No. 2689/30015.

----------END----------

☐ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

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