

HORTON AVIATION LLC.



**AIRFRAME
MAINTENANCE
RECORDS**



AIRFRAME MAINTENANCE RECORDS

Log No. _____

Aircraft Registration No. N4542T

Aircraft Mfg. Piper Model PA34-200 Serial No. 347250128

Engine Mfg. _____ Model _____ Serial No. _____

Engine Mfg. _____ Model _____ Serial No. _____

Propeller Mfg. _____ Model _____ Serial No. _____

Hub Design No. _____ Hub Serial No. _____

Blade Design No. _____ Blade Serial No. _____

Propeller Mfg. _____ Model _____ Serial No. _____

Hub Design No. _____ Hub Serial No. _____

Blade Design No. _____ Blade Serial No. _____

DESCRIPTION OF WORK PERFORMED
SIGNATURE & CERTIFICATE NO. OF PERSON PERFORMING WORK

TOTALS brought forward from previous page

HORTON AVIATION SERVICES LLC
2732 PERIMETER ROAD, SUITE 101
NORTH LAS VEGAS, NEVADA 89032
REPAIR STATION # YHSR284L

N# N4542T MFG. PIPER MODEL PA34-200 S/N 34-7250128 W.O.# 07-231 DATE 08/01/08 HOBBS 2378.4

AD 73-11-02, AD 75-24-02, AD 2002-26-01, AD 2005-13-16 COMPLIED WITH. REFERENCE AD COMPLIANCE LOG THIS DAY. 1) REMOVED AND REPLACED BATTERY, P/N G-35, 2) REPLACED A/C TRIM SCREWS WITH NEW STAINLESS STEEL KIT, P/N SENECA TRIM KIT, 3) REMOVED AND REPLACED ELT, P/N ON ME406, 4) REPLACED FUEL STRAINER GASKETS, 5) REMOVED AND REPLACED MAIN CABIN DOOR ROD ASSEMBLY, 6) REMOVED AND REPLACED SEAT BELTS, 7) INSPECTION OF AFFECTED AREA REVEALED PRIOR RE-INFORCEMENT BRACKETS INSTALLED. APPEARS WORK COMPILED WITH BEFORE SB 1161 ISSUED. 8) REPAIRED NOSE BAGGAGE CRACKS AND MISSING HARDWARE, 9) REMOVED AND REPLACED STABILATOR, REPLACED ASSEMBLY BUSHINGS TORQUE TO SPEC 40 IN. LBS. CHECKED STABILIZER UP AND DOWN, 10) REMOVED AND REINSTALLED BALANCE TUBE AFTER TREATED WITH ANTI CORROSIVE ZINC CLEANED AND INSPECT FOR CRACKS, 11) REPLACED AND CLEANED LH & RH FUEL CAP, INSTALLED NEW GASKETS, 12) REMOVED AND REPLACED MAIN VACUUM FILTER D9-18-1 DUE TO TIME LIMITS, 13) REMOVED RUDDER AND TREATED CORROSION AT AND AROUND THE BASE. APPLIED ZINC CHROMATE AT AND AROUND THE BASE OF THE RUDDER. REINSTALLED RUDDER IAW PIPER PA-34-200MM, 14) REPAIRED RIGHT MAIN STRUT, REPLACE O-RINGS, SCRAPER RINGS, SERVICE W/5606 HYDRAULIC FLUIDS, 15) INSTALLED ALL NEW DUCTING BEHIND INSTRUMENT PANEL, 16) REMOVED AND REPLACED FLAT SPOTTED NOSE TIRE AND TUBE. SERVICED TIRE TO 31 psi IAW PIPER PA-34-200 MM. 17) REMOVED AND REPLACED BOTH L/H AND R/H BRAKE CALIPERS, 18) REPLACED LANDING GEAR ACTUATOR FLEX LINES AT ALL 3 GEARS, 19) HYDRO-CHECK OXYGEN BOTTLE DUE TO TIME LIMITS, 20) REPLACED UPPER AND LOWER MAIN TORQUE LINKS BUSHINGS AND BOLTS. SAFETY AND LOCK NUTS IAW PIPER 34-200 MM, 21) REPLACED RIGHT MAIN GEAR SIDE BRACE STUD WITH NEW ASSY, 22) REMOVED AND REPLACED SHIMMY DAMPNER USING ALL NEW HARDWARE, 23) REPLACED STEERING ROLLER AND BOLTS, 24) REMOVED AND REPLACED WIRING HARNESS, AND UPHOLSTERY, INSTALLED HYDRAULIC JACK NOSE GEAR KIT, 25) REMOVED FORCE AIR AND INSTALLED AVIONICS COOLING FAN, 26) REMOVED AND REPLACED RIGHT WING HEATER INLET DUCT WITH NEW SCHEET. 27) REPAIRED MASTER CYLINDER ON RH OUTBRD AND INBRD, 28) FABRICATED AND INSTALLED BRACKET FOR NAV RS-08 SWITCHING MODULES, 29) FABRICATED MOUNTING BRACKET AND INSTALLED AVIONIC MASTER RELAY IN R.H. SIDE OF A/C FWD OF INSTRUMENT PANEL, 30) REPLACED AND REROUTED HEATING DUCTS, 31) REPLACED ANTI SQUEAK WEBBING ON BOTH COWLING, 32) REPLACED LEFT DOWN LIMIT GEAR MICRO SWITCH, 33) REMOVED AND REROUTED OXYGEN LINES, 34) REMOVED AND REPLACED BRAKE HOSES, 35) REROUTED STATIC LINES, 36) REMOVED AND REPLACED FRONT WIND SHIELD IAW STC, 37) REROUTED FLAP WIRES, 38) REROUTED DOOR SEAL HOSE, 39) REMOVED AND REPLACED SIDE WINDOWS IAW STC, 40) ADDED ALTIMETER TO CO-PILOT'S INSTRUMENT PANEL, 41) REMOVE AND REPLACED HOBBS HOURS WHEN REPLACED 2378.4, 42) INSTALLED VORTEX GENERATORS IAW STC, 43) REMOVED AND REPLACED A/C DOOR LOCKS, 44) relocated compass and compass swing c/w, 45) REMOVED REFACED AND REINSTALLED PILOT'S VSI AND AIRSPEED INDICATOR, 46) REMOVED WING TIPS AND INSTALLED LOPRESTI WING TIPS IAW STC, 47) REMOVED AND REPLACED GEAR INDICATION LAMPS, 48) REMOVED AND REPLACED TRIM CABLE, 49) REMOVED THROTTLE HOUSING AND REPAIR CRACKS WITH FIBER GLASS, 50) REMOVED AND REPLACED LEFT COWL FLAP CABLE, 51) 91.411 AND 91.413 CHECK COMPLIED WITH SYSTEM TESTED AND FOUND TO COMPLY WITH F.A.R. 43 APPENDIX E AND F, 52) I certify that this aircraft has been inspected in accordance with an annual inspection and was determined to be in airworthy condition.

[Signature]


COMPASS swing N=358, 30=30, 60=60, 90=93

120=126, 150=154, 180=182, 210=212, 240=240, 270=269, 300=300, 330=330

SUB TOTALS this page

[Signature]
YHSR284L


TOTALS- carry forward to next page

DATE	TOTAL TIME IN SERVICE	TOTAL TIME SINCE OVERHAUL	TACH OR RECORDING METER TIME	DESCRIPTION OF WORK PERFORMED SIGNATURE & CERTIFICATE NO. OF PERSON PERFORMING WORK
TOTALS brought forward from previous page				
HORTON AVIATION SERVICES LLC 2732 PERIMETER ROAD, SUITE 101 NORTH LAS VEGAS, NEVADA 89032 REPAIR STATION # YHSR284L				
N# N4542t MFG. PIPER MODEL PA34-200T S/N 34-7250128 W.O.# 07-231 DATE 08/03/08 HOBBS 2378.4 TROUBLESHOT LANDING GEAR SYSTEM REMOVED AND REPLACED DOWN LIMIT SWITCHES AND UP LIMIT SWITCHES IN BOTH RIGHT MAIN AND LEFT MAIN LANDING GEAR, AND IN NOSE GEAR. ALL SYSTEMS OPERATION CHECKED GOOD, APPROVED FOR RETURN TO SERVICE. 08/22/2008. JAMES D. HORTON 				

HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA34-200 S/N 34-7250128 W.O. 09-006 DATE 01/24/2009 TACH. 2983.96

REPLACED STARTER ON RIGHT ENGINE. RESEALED RIGHT GEAR ACTUATOR AND BLEED SYSTEM. REPLACED RIGHT MASTER CYLINDER ON PILOT SIDE WITH NEW CLEVELAND MASTER CYLINDER AND BLEED BRAKE SYSTEM. FUNCTIONAL CHECK OF SYSTEMS NORMAL.

The aircraft and/or component above were repaired and inspected in accordance with Federal Aviation Regulations and are approved for return to service. Pertinent details are on file at this agency under above work order number. APPROVED FOR RETURN TO SERVICE


April 17, 2009; Piper PA-34-200; Serial #34-7250128; N4542T; 1972 Year Model; Hobbs 0089.5


Removed flight controls for painting. Completely strip etch, alodine and paint aircraft. Removed and replace flap rollers. Checked balance of control surfaces after painting, in accordance with A/C services manual. All static moments determined to be within limits, re-install controls.


ED FLEET 229761220 AP

April 17, 2009; Piper PA-34-200; Serial #34-7250128; N4542T; 1972 Year Model; Hobbs 0089.5

PAINT COLORS:

FIRST STAR : 570760; Jet Glo
SKY BLUE MET: H10499@; Acry Glo
LIGHT RICH BLUE MET: H10503@; Acry Glo
LAS VEGAS GOLD MET: H10494@; Acry Glo


Gustav A. Haussler III, Treasurer
Master Aircraft Service, Inc.
3475 N. Sabin Brown Road
Wickenburg, AZ 853903
928+684-4926

SUB TOTALS this page

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HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA34-0200 S/N 34-7250128 W.O. 09-088 DATE 04/21/2009 TACH. 2997.59

1 Removed old squat switch, installed new squat switch IAW Piper MM. All hardware torque to spec system ops check good. 2 Removed autopilot for factory repair. Reinstalled auto pilot after return from factory. 3 Replaced right main tire with new Goodyear 600X6 Flight Custom.

The aircraft and/or component above were repaired and inspected in accordance with Federal Aviation Regulations and are approved for return to service. Pertinent details are on file at this agency under above work order number. APPROVED FOR RETURN TO SERVICE

HORTON AVIATION SERVICES LLC
2732 PERIMETER ROAD, SUITE 101
NORTH LAS VEGAS, NEVADA 89032
REPAIR STATION # YHSR284L

N# N4542T MFG. PIPER MODEL PA34-200 S/N 34-7250128 W.O.# 09-151 DATE 08/15/09 HOBBS 111.0 AFTT: 5349.2

Replaced all pulleys in roll control system, Rigged roll control system I/A/W Robertson service and maintenance manual SMM 27-1, Resealed hand brake cylinder and bled complete brake system, replaced left torque plate, Rerigged cabin door latch assy. to prevent sticking on lower door, Replaced left main tire and tube due to flat spot, Replaced throttle and mixture controls for left and right engines and rigged I/A/W manufactures maintenance manual, Installed new bumpers in main gear wheel well, Replaced turn coordinator, reference attached 8130 for details, Installed new springs in cowl flap control levers, ELT inspected I/A/W FAR 91.207 (d) and found to be in satisfactory condition. Reference attached AD compliance sheets for AD details. I certify that this airframe has been inspected I/A/W an annual inspection and determined to be in an airworthy condition.

HORTON AVIATION SERVICES LLC
2732 PERIMETER ROAD, SUITE 101
NORTH LAS VEGAS, NEVADA 89032
REPAIR STATION # YHSR284L

N# N4542T MFG. PIPER MODEL PA34-200T S/N 34-7250128 W.O.# 10-154 DATE 08/06/2010 HOBBS 3042.0

TROUBLESHOT LANDING GEAR SYSTEM, FOUND LEFT ACTUATOR LEAKING, REMOVED UNIT FROM AIRCRAFT, REMOVED AND REPLACED ALL O-RINGS AND SEALS, REINSTALLED UNIT ON AIRCRAFT, REFILLED HYDRAULIC, BLEED SYSTEM, OPERATION CHECK SYSTEM GOOD.
The aircraft and/or component above were repaired and inspected in accordance with Federal Aviation Regulations and are approved for return to service. Pertinent details are on file at this agency under above work order number. APPROVED FOR RETURN TO SERVICE.

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				TOTALS brought forward from previous page

HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128 W.O.11-038 Date: 02/12/2011 Left Tach
Time: 3053.31 Right Tach: 3047.26 Hobbs: 176.8 Total Time: 5415.0

Removed and replaced induction air filters. Repaired left cowl flap control attach point. Removed left flap control rod attach bolts installed spacers as needed and reinstalled bolts. C/W AD 73-11-02, AD75-24-02, AD2010-15-10, AD 2008-14-07, 2009-22-03, AD2008-13-28 Reference AD compliance list for details. ELT inspected I/A/W FAR 91.207 (d) and found to be in satisfactory condition. I certify that this airframe has been inspected in accordance with an annual inspection and was determined to be in an airworthy condition. OPERATION CHECKED SYSTEMS GOOD. APPROVED FOR RETURN TO SERVICE. *Charles*

HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128 W.O.11-038 Date: 02/12/2011 Left Tach
Time: 3053.31 Right Tach: 3047.26 Hobbs: 176.8 Total Time: 5415.0

Removed and replaced main gear door hinges and rigged gear doors I/A/W manufactures maintenance manual. OPERATION CHECKED SYSTEMS GOOD. APPROVED FOR RETURN TO SERVICE. *Charles*

HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N4542T MFG. PIPER Model: PA32-200 S/N: 347250128 W.O.:11-038 Date:2/12/2011 LEFT TACH:3053.31
RIGHT TACH: 3047.26 HOBBS 176.8 TOTAL TIME: 5415.0

ALTIMETER(S) TESTED TO 20,000FT
 I CERTIFY THAT THE STATIC SYSTEM(S), ALTIMETER(S), AND AUTOMATIC PRESSURE ALTITUDE REPORTING SYSTEM(S) WAS TESTED AND INSPECTED AS REQUIRED BY FAR 91:411 AND FOUND TO COMPLY WITH FAR 43, APPENDIX E AND F. I CERTIFY THAT THE ATC TRANSPONDER WAS TESTED AND INSPECTED AS REQUIRED BY FAR 91:413 AND FOUND TO COMPLY WITH FAR 43, APPENDIX F.

The aircraft and/or component above was repaired and inspected in accordance with Federal Aviation Regulations and is approved for return to service. Pertinent details are on file at this agency under above work order number. *James D. Horton*

HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128 W.O.11-105 Date: 05/12/2011 Left Tach
Time: 3053.31 Right Tach: 3047.26 Hobbs: 176.8 Total Time: 5415.0

Removed and replaced turn coordinator, IAW manufactures maintenance manual. P/N on 1394T100-14RB, S/N on A07-10971. OPERATION CHECKED SYSTEMS GOOD. APPROVED FOR RETURN TO SERVICE. *James D. Horton*

SUB TOTALS this page

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HORTON AVIATION SERVICES LLC
2732 PERIMETER ROAD, SUITE 101
NORTH LAS VEGAS, NEVADA 89032
REPAIR STATION # YHSR284L

N# N4542T MFG. PIPER MODEL: PA34-200 S/N: 347250128
W.O.#13-202 DATE: 9/1/2013 HOBBS: 3116.75

ALTIMETER(S) TESTED TO 20,000 feet,

I CERTIFY THAT THE STATIC SYSTEM, ALTIMETER, AIR DATA
COMPUTER, AND AUTOMATIC PRESSURE ALTITUDE
REPORTING SYSTEM(S) WAS TESTED AND INSPECTED AS
REQUIRED BY FAR 91.411 AND FOUND TO COMPLY WITH FAR
43, APPENDIX E AND F.

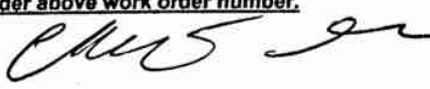
I CERTIFY THAT THE ATC TRANSPONDERS, WAS TESTED AND
INSPECTED AS REQUIRED BY FAR 91.413 AND FOUND TO
COMPLY WITH FAR 43, APPENDIX F.



HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128 W.O.16-
008 Date: 2/19/2016 Left Tach Time: 3116.23 Right Tach: 3111.29
Hobbs: 347.9 Total Time: 5477.74

Replaced induction air filters. Resealed right wheel cylinder and replaced brake
linings. Removed and reinstalled O2 bottle after hydrostatic inspection. Reattached
door seal on forward baggage door. Replaced aircraft battery. Installed missing
spacers on center seat belts. Replaced ELT battery. Replaced all hoses on both
engines. CW AD 73-11-02, CW AD 2015-19-07 left engine, CW AD 2015-19-07
right engine< CW AD 2008-13-28 right prop, CW AD 81-19-04 left engine, CW AD
31-19-04 right engine, CW AD 75-24-02 reference AD compliance sheet for details.
Elt inspected I/AW FAR 91.207 (d) and found to be in satisfactory condition. I
certify that this airframe has been inspected in accordance with an annual
inspection and was determined to be in an airworthy condition. The aircraft and/or
components above was repaired and inspected in accordance with Federal
Aviation Regulations and is approved for return to service. Pertinent details
are on file at this agency under above work order number.



HORTON AVIATION SERVICES

2732 Perimeter Rd Ste. 101

N. Las Vegas, NV 89030

FAA Approved Repair Station #YHSR-284L

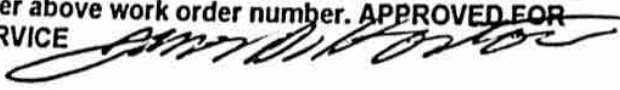
N# N4542T Mfg. PIPER Model PA34-200T

S/N 34-7250128 W.O.# 16-008 Date 02/26/2016

TACH:left 3118.02; right 3111.41

I certify that the ATC transponder(s) was tested and inspected as
required by FAR 91.413 and found to comply with FAR 43, Apdx. F

The aircraft and/or component above were repaired and
inspected in accordance with Federal Aviation Regulations and
are approved for return to service. Pertinent details are on file at
this agency under above work order number. **APPROVED FOR
RETURN TO SERVICE**



HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O.17-059 Date: 3/17/2017 Left Tach Time: 3161.22 Right Tach:
3154.19 Hobbs: 400.1 Total Time: 5522.73

Replaced induction air filters. Replaced fuel drain valves on left and
right fuel sumps. CW AD 73-11-02, CW AD 2015-19-07 left engine,
CW AD 2015-19-07 right engine< CW AD 2008-13-28 right prop, ,
CW AD 75-24-02 reference AD compliance sheet for details. Elt
inspected I/AW FAR 91.207 (d) and found to be in satisfactory
condition. I certify that this airframe has been inspected in
accordance with an annual inspection and was determined to be in
an airworthy condition.

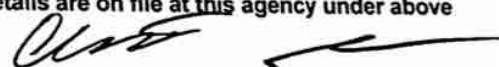
The aircraft and/or components above was repaired and
inspected in accordance with Federal Aviation Regulations and
is approved for return to service. Pertinent details are on file at
this agency under above work order number.



HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O.18-148 Date: 6/1/2018 Left Tach Time: 3185.3 Right Tach:
3178.0 Hobbs: 428.4 Total Time: 5546.81

Replaced induction air filters. Repaired broken wire at flap switches.
Replaced right engine #4 EGT probe. CW AD 73-11-02, CW AD
2015-19-07 left engine, CW AD 2015-19-07 right engine< CW AD
2008-13-28 right prop, , CW AD 75-24-02 reference AD compliance
sheet for details. Elt inspected I/AW FAR 91.207 (d) and found to be
in satisfactory condition. I certify that this airframe has been
inspected in accordance with an annual inspection and was
determined to be in an airworthy condition. The aircraft and/or
components above was repaired and inspected in accordance
with Federal Aviation Regulations and is approved for return to
service. Pertinent details are on file at this agency under above
work order number.



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HORTON AVIATION SERVICES
2732 Perimeter Rd. Ste. 101
N. Las Vegas, NV 89030
FAA Approved Repair Station #YHSR-284L

N# N4542T Mfg. PIPER Model PA34-200T
S/N 34-7250128 W.O.# 16-008 Date 02/26/2016
TACH: left 3118.02; 3111.41

I certify that the static system(s), altimeter(s) and automatic pressure altitude reporting system(s) was tested and inspected as required by FAR 91.411 and found to comply with FAR 43, Appendix E and F. Altimeter(s) tested to 20,000 FEET.

The aircraft and/or component above were repaired and inspected in accordance with Federal Aviation Regulations and are approved for return to service. Pertinent details are on file at this agency under above work order number. **APPROVED FOR RETURN TO SERVICE**

[Signature]

HORTON AVIATION SERVICES LLC

FAA CRS YHSR-284L

2732 PERIMETER ROAD, SUITE 101

N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O.16-172 Date: 4/16/2016 Left Tach Time: 3132.08 Right Tach: 3125.32 Hobbs: 366.7

Inspected and bled left brake system.

The aircraft and/or components above was repaired and inspected in accordance with Federal Aviation Regulations and is approved for return to service. Pertinent details are on file at this agency under above work order number.

[Signature]

HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O.17-290 Date: 12/02/2017 Left Tach Time: 3173.92 Right Tach: 3166.77 Hobbs: 415.03

Troubleshoot charging system and replaced right alternator and both voltage regulators. Test ran aircraft and adjusted regulators I/A/W Piper services manual. The aircraft and/or components above was repaired and inspected in accordance with Federal Aviation Regulations and is approved for return to service. Pertinent details are on file at this agency under above work order number.

[Signature]

DESCRIPTION OF WORK PERFORMED
NATURE & CERTIFICATE NO. OF PERSON PERFORMING WORK

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HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O.16-062 Date: 3/11/2016 Left Tach Time: 3122.13 Right Tach: 3115.44 Hobbs: 355.2

Removed and replaced all fuel tank sumps.

The aircraft and/or components above was repaired and inspected in accordance with Federal Aviation Regulations and is approved for return to service. Pertinent details are on file at this agency under above work order number.

[Signature]

HORTON AVIATION SERVICES LLC
FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101
N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O. #18-011 DATE 1/9/2018 Left Tach: 3177.9 LEFT ENGINE IO-360-C1E6 S/N L-8596-51A

Removed and replaced vacuum pump. Removed and replaced vacuum pump drive pad gasket and seal. Test run and leak check normal. The aircraft and/or components above was repaired and inspected in accordance with Federal Aviation Regulations and is approved for return to service. Pertinent details are on file at this agency under above work order number.

[Signature]

HORTON AVIATION SERVICES LLC FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101 N. LAS VEGAS NV 89032
N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O.19-190 Date: 9/9/2019 Left Tach Time: 3228.16 Right Tach: 3220.59 Hobbs: 477.3 Total Time: 5595.71
C/W AD 75-24-02, CW AD 73-11-02, CW AD 2008-13-28 right prop, CW AD 2011-26-04, CW AD 2009-22-03, CW AD 2008-13-28, CW AD 2009-22-03, reference AD compliance sheet for details. Secured spark plug leads, replaced fitting on pilot's left cylinder, swung compass and installed new compass card, ELT inspected I/A/W FAR 91.207 (d) and found to be in satisfactory condition. I certify that this airframe has been inspected in accordance with an annual inspection and was determined to be in an airworthy condition.

[Signature] 339815LW

HORTON AVIATION SERVICES LLC FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101 N. LAS VEGAS NV 89032
N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O.20-172 Date: 10/23/2020 Left Tach Time: 3257.83 Right Tach: 3250.03 Hobbs: 518.2 Total Time: 5625.15
ALTIMETER TESTED TO 20,000 feet, I CERTIFY THAT THE STATIC SYSTEM, ALTIMETER, AND AUTOMATIC PRESSURE ALTITUDE REPORTING SYSTEM WAS TESTED AND INSPECTED AS REQUIRED BY FAR 91.411 AND FOUND TO COMPLY WITH FAR 43, APPENDIX E AND F. The aircraft and/or component above were repaired and inspected in accordance with Federal Aviation Regulations and are approved for return to service. Pertinent details are on file at this agency under above work order number. **APPROVED FOR RETURN TO SERVICE**

[Signature]

DATE	TOTAL TIME IN SERVICE	TOTAL TIME SINCE OVERHAUL	TACH OR RECORDING METER TIME	SIG
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HORTON AVIATION SERVICES LLC FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101 N. LAS VEGAS NV 89032
N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O.20-172 Date: 10/23/2023 Left Tach Time: 3257.83 Right
Tach: 3250.03 Hobbs: 518.2 Total Time: 5625.15
I CERTIFY THAT THE ATC TRANSPONDERS WAS TESTED AND
INSPECTED AS REQUIRED BY FAR 91.413 AND FOUND TO
COMPLY WITH FAR 43, APPENDIX F.
The aircraft and/or component above was repaired and
inspected in accordance with Federal Aviation Regulations and
is approved for return to service. Pertinent details are on file at
this agency under above work order number.

HORTON AVIATION SERVICES LLC FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101 N. LAS VEGAS NV 89032
N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O.20-127 Date: 10/22/2020 Left Tach Time: 3257.83 Right
Tach: 3250.03 Hobbs: 518.2 Total Time: 5625.15 C/W AD 75-24-
02, C/W AD 73-11-02, C/W AD 2008-13-28 right prop, C/W AD 2015-
02, C/W AD 2004-10-14, C/W AD 2008-13-28, C/W AD 81-19-04
19-07, C/W AD 2004-10-14, C/W AD 2005-13-16, C/W AD 2009-22-03,
C/W AD 2013-02-13, C/W AD 2005-13-16, C/W AD 2009-22-03,
reference AD compliance sheet for details. Removed vacuum filter,
replaced both alternator circuit breakers, Secured rudder tab, and
throttle and mixture cables, replaced right main tube, cleaned
battery cables, replaced both air filters, replaced fuel cap gaskets,
hydro test oxygen bottle, resealed right and left main struts, replaced
gaskets on #3 cylinder intake, spark plug leads, replaced fitting on
pilot's left cylinder, ELT inspected I/A/W FAR 91.207 (d) and found
to be in satisfactory condition. As required by F.A.R. 91.409 I certify
that this airframe's Annual inspection was complied with I/A/W
F.A.R. 43 Appendix D (Scope and Detail of Items To Be Included in
Annual and 100-Hour Inspections), and found to be in an airworthy
condition. The aircraft and/or components above was repaired
and inspected in accordance with Federal Aviation Regulations
and is approved for return to service. Pertinent details are on
file at this agency under above work order number.

HORTON AVIATION SERVICES LLC FAA CRS YHSR-284L
2732 PERIMETER ROAD, SUITE 101 N. LAS VEGAS NV 89032

N# N4542T MFG. PIPER MODEL PA-34-200 S/N 347250128
W.O.20-236 Date: 12/22/2020 Left Tach Time: 3257.85 Right
Tach: 3250.05 Hobbs: 518.8

Removed Sandel EHSI SN3500-002, and KG102A remote gyro,
and KMT 112 Azimuth, and Vertical speed indicator, Installed
new dual G5 indicators in pilot's instrument panel I.A.W. STC
SA01818WI, weight and balance and equipment list revised. The
aircraft and/or components above was repaired and inspected
in accordance with Federal Aviation Regulations and is
approved for return to service. Pertinent details are on file at
this agency under above work order number.

HORTON AVIATION SERVICES LLC 2732
PERIMETER ROAD, SUITE 101 NORTH LAS
VEGAS, NEVADA 89032 REPAIR STATION #
YHSR284L N# N4542T MFG. PIPER MODEL PA34-
200T S/N 34-7250128 W.O.# 20-252 DATE
12/17/2020 LEFT TACH: 3263.57, LEFT ENGINE
IO-360-C1E6 S/N L-8596-51A

REPAIRED BROKEN HINGE ON LEFT
FORWARD AFT ON LEFT COWLING ON THE
LEFT ENGINE. FABRICATED TWO
REINFORCEMENT BRACKETS AND SECURED
TO HINGE. OPS CHECK GOOD The aircraft
and/or components above was repaired and
inspected in accordance with Federal Aviation
Regulations and is approved for return to
service. Pertinent details are on file at this
agency under above work order number.

SUB TOTALS this page

TOTALS- carry forward to next page

Wt_Bal


EQUIPMENT CHANGE - WEIGHT & BALANCE		HORTON AVIATION SERVICES LLC	
REG. NO. N4542T	MODEL PA34-200	Serial No. 34-7250128	
Items: (Description / P/N / S/N)	Weight Pounds	Arm Inches	Moments Inch/Pounds
Previous Aircraft Empty Weight:	3151	87.33	275163.6
REMOVED; VERTICAL SPEED IND.	-1	65	-65
KMT 112 FLUX DETECTOR	-0.5	232	-116
KG 102A REMOTE GYRO	-5	202	-1010
AK550-6 POWER SUPPLY	-2.3	40	-92
SN3500 H.S.I.	-3	63	-189
ADDED;	0	0	0
DUAL G5 INDICATORS	1.96	65	127.4
GAD 29B	0.63	36	22.68
GMU 11	0.26	110	28.6
	0	0	0
	0	0	0
	0	0	0
	0	0	0
	0	0	0
	0	0	0
Totals	3142.05		273870.28

A. Old Empty Weight	3151	Pounds
B. Old Empty CG	87.33	Inches
C. Old Empty Weight CG Moment	275163.6	Inch/Pounds
D. Max Gross Weight	4200	Pounds
E. Old Useful Load	1049	Pounds

A. New Empty Weight	3142.05	Pounds
B. New Empty CG	87.162929	Inches
C. New Empty Weight CG Moment	273870.28	Inch/Pounds
D. Max Gross Weight	4200	Pounds
E. New Useful Load	1057.95	Pounds

This new weight & balance information superseads all previous weight and balance data.
 For aircraft loading, see instructions in Weight & Balance Section of Aircraft Flight Manual.

FAA Form 337 Completed?	N/A
Equipment List Amended?	12/7/2020

BY: JAMES HORTON	Date:	12/7/2020
FAA REPAIR STATION YHSR284L		
SIGNATURE: 		
NOTES LEVEL ANALYSIS ON STC SA00118MC REDUCES MTOW TO 3750		

EQUIPMENT LIST

FAA APPROVED REPAIR STATION NO; YHSR284L

HORTON AVIATION SERVICES LLC

2732 PERIMETER RD. STE. 101

N. LAS VEGAS, NV. 89032

N# N4542T A/C MFG. PIPER MODEL PA34-200 SERIAL NO. 34-7250128

ITEM			
AIRCRAFT EMPTY			
<u>NOUN</u>	<u>S/N</u>	<u>WEIGHT</u>	<u>ARM</u>
GI 106A INDICATOR	B08-10334	1.25	64.9
ELT ME406		2.5	262.0
ELT REMOTE SWITCH	345-6196-04	0.2	66.0
AIR SPEED INDICATOR	24618	1.0	65.0
FLIGHT DIRECTOR	1170	1.5	65.0
GDL 69A DATA LINK	47759191	2.0	202.0
TAS 600 PROCESSOR	0711210	5.5	202.0
PMA8000B AUDIO PANEL	S03463	1.4	64.0
GNS 530W GPS/COM/NAV	23802713	9.0	62.0
SYSTEM 55X AUTO PILOT	0801-13113	2.7	62.0
GTX 327 TRANSPONDER	83731263	3.1	64.0
GNS 430W GPS/COM/NAV	23404309	7.0	62.0
EDM-760 MONITOR	5-2004	3.0	63.0
TURN COORDINATOR	L07-11598	1.4	64.0
COMPASS	10873	0.6	66.0
ENCODER SSD120-RS232	SRA-13260	1.0	61.0
ENCODING / ALTIMETER	127122	2.0	63.0
MANIFOLD PRESSURE IND	53249	1.0	63.0
SHADIN FUEL FLOW	2306	1.0	63.0
RS08 NAV SWITCH	11382	1.0	36.0
RS08 NAV SWITCH	3848	1.0	36.0
ALTIMETER	5G855	1.1	64.0
HOBBS HOUR METER	85331	0.3	65.0
DUAL G5 INDICATORS	4JQ044974/4JQ044949	1.96	65.0
GAD 29B	5DL012790	.63	36.0
GMU 11	56J015109	.26	110

DATE: 12/07/2020 REVISED

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N4542T

Nationality and Registration Mark

12/05/2020

Date

1. *Installed dual Garmin G5 indicators in pilot's instrument panel IAW STC SA01818WI. Instructions for continued airworthiness provided with AFM, and kept in aircraft records. Aircraft weight and balance and equipment list have been updated.*

United States of America
Department of Transportation — Federal Aviation Administration
Supplemental Type Certificate

Number SA900EA

This certificate, issued to

B. F. Goodrich Aerospace and Defense Products
A Division of the B. F. Goodrich Company
500 South Main Street
Akron, Ohio 44318

*certifies that the change in the type design for the following product with the limitations and conditions
therefor as specified hereon meets the airworthiness requirements of Part 23 of the Federal Aviation
Regulations.*

Original Product — Type Certificate Number: A750

Make: Piper

Model: PA34-200

Description of Type Design Change:

Installation of B. F. Goodrich Electrothermal Propeller De-Icer System Kit
No. 77-220 in accordance with B. F. Goodrich Report No.'s 70-04-700 dated
1 May 1970, 71-04-717B dated 2 March 1972 and 59-728E dated 1 August 1970.

Limitations and Conditions:

1. This STC is applicable to S/N's 34-7250001 and subsequent.
2. B. F. Goodrich Company Airplane Flight Manual Supplement No. 1, FAA Approved 18 January 1972 to the Piper PA34-200 Flight Manual is required with this installation.

(See STC Continuation Sheet Page 2)

*This certificate and the supporting data which is the basis for approval shall remain in effect until sur-
rendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the
Federal Aviation Administration.*

Date of application: 4 October 1971

Date reissued:

Date of issuance: 15 March 1972

Date amended: 10 April 1972



By direction of the Administrator

for W. O. Lebsack
W. F. NORTON

(Signature)

Chief, Engineering and Manufacturing Branch
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

United States of America
Department of Transportation—Federal Aviation Administration
Supplemental Type Certificate

Number SA01681SE

This certificate, issued to

Micro AeroDynamics, Inc.
4000 Airport Road, Suite D
Anacortes, WA 98221

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 23 of the Federal Aviation Regulations.

Original Product—Type Certificate Number:

*See attached FAA Approved Model List (AML)
No. SA01681SE for a list of approved airplane
models and applicable airworthiness regulations

Make:

Model:

Description of the Type Design Change: Installation of vortex generators on the wings, horizontal and vertical tail surfaces (all models), and an increase in maximum zero fuel weight (PA-34-200T and PZL M20 03 only) in accordance with FAA-approved Micro AeroDynamics, Inc. Drawing Package MA2212, Revision A, dated August 14, 2006, and Installation Manual MA2213, Revision A, dated October 6, 2006, or later FAA-approved revisions to these documents.

NOTE: If more than five vortex generators are damaged or missing, the aircraft is not airworthy. The increase in maximum zero fuel weight (PA-34-200T and PZL M20 03 only) is also applicable to installations done in accordance with Installation Manual MA2213, Revision IR dated May 15, 2006

Limitations and Conditions: Approval of this change in type design applies to the above aircraft only. This approval should not be extended to other aircraft on which other previously approved modifications are incorporated unless it is determined that the relationship between this change and any other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that aircraft. A copy of this Certificate and FAA-Approved Flight Manual Supplement MA2215 Revision IR dated October 25, 2006 (PA-34-200T and PZL M20 03 only) must be maintained as part of the permanent records for the modified aircraft.

If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

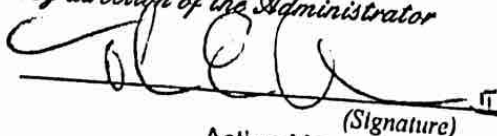
Date of application: March 17, 2006

Date of issuance: June 16, 2006

Date reissued:

Date amended: October 25, 2006

By direction of the Administrator



(Signature)

Acting Manager, Seattle Aircraft
Certification Office
(Title)



Alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FORM 8110-2(10-68)

This certificate may be transferred.

Life is Short ... Fly **FAST!**



Speed Merchants
by
LoPresti

Master Drawing List

Report No. 33

Constant Chord Piper Wing Tip

PA28-140/150/160/180
PA28R-180/R200
PA34 Series

Number of Pages 1 through 3 inclusive

Rev. F
06/07/06

Prepared By: _____

Curt LoPresti

Approved By: _____

Curt LoPresti

LoPresti Speed Merchants
800-859-4757 / 772-562-4757 or Fax 772-563-0446
<http://www.Speedmods.com>
2620 Airport North Drive, Vero Beach, Florida, 32960 U.S.A.

United States of America
Department of Transportation -- Federal Aviation Administration

Supplemental Type Certificate

Number SA02279AT

This certificate issued to LoPresti Speed Merchants
2620 Airport North Drive
Vero Beach, Florida 32960

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified herein meets the airworthiness requirements of Part 21 of the Federal Aviation Regulations.

Original Product: Type Certificate Number: * See attached

Make: FAA Approved Model List (AML)

Model: Document LSM-500-025 for a list of
Approved Airplane Models

Description of Type Design Change:

Installation of a Boom Beam Bulb, Starter and Ballast in accordance with Master Drawing List Report Number 43, Revision F, dated 15 July 2002, or later FAA Approved Revision.

Limitations and Conditions:

This approval should not be extended to other aircraft of this model on which other previously approved modifications are incorporated unless it is determined by the installer that the interrelationship between this change and any of those other previously approved modifications will produce no adverse effect upon the airworthiness of that airplane. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval of this modification shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: August 15, 2002

Date of issuance: February 14, 2001



Authorization for the application of this
modification is given by LSM for use on
aircraft number

Amended: April 5, 2001; July 22, 2002;
February 14, 2003

By direction of the Administrator

Eugene R. Bollini
(Signature)

for Melvin D. Taylor, Manager,
Atlanta Aircraft Certification Office

(Date)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 1 year, or both.
This certificate may be transferred to another person with the consent of the Administrator.

United States Of America
Department of Transportation - Federal Aviation Administration

Supplemental Type Certificate

Number SA8897SW-D

This Certificate issued to S-TEC Corporation
One S-TEC Way
Mineral Wells Municipal Airport
Mineral Wells, TX 76067-9236

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 23 of the Federal Aviation Regulations.

Original Product Type Certificate Number: A7SO

Make: Piper

Model: PA-34-200, PA-34-200T, and PA-34-220T

Description of Type Design Change:

Installation of S-TEC System 55/55X Two Axis Automatic Flight Guidance System, Model ST-567, according to Bulletin No. 667, Revision 5, dated 5-16-01, and Master Drawing List No. 92736, Revision E, dated 5-16-01, or later FAA Approved revisions of the above data (14 Volt System).

Limitations and Conditions:

(See Continuation Sheet, Page 2, a part of this STC.)

If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: 12-08-94

Date of issuance: 12-28-94

Date reissued:

Date amended: 5-16-01

By direction of the Administrator



(Signature)

Michael Keirman
DAS Staff Coordinator, DAS 5 SW

(Title)



U.S. Department
of Transportation
Federal Aviation
Administration

TRANSPORT AIRPLANE DIRECTORATE
AIRCRAFT CERTIFICATION SERVICE
LOS ANGELES AIRCRAFT CERTIFICATION OFFICE
3960 PARAMOUNT BOULEVARD
LAKEWOOD, CALIFORNIA 90712-4137

JUL 08 1996

J. P. Instruments
3402-1 West Mac Arthur
Santa Ana, CA 92704

Dear Mr. Polizzotto:

J. P. Instruments, Temperature Indicator,
Technical Standard Order C43c

Your application dated March 28, 1996, requesting the issuance of a Technical Standard Order (TSO) authorization in accordance with the procedural requirements of Federal Aviation Regulation (FAR) Part 21, Subpart O has been reviewed. Based upon your data and statement of conformance certifying your article(s) have met the requirements of FAR Part 21, Subpart O, and the minimum performance standards of TSO C43c (Ref. FAR 21.305(b)), authorization is hereby granted for the following:

MODEL	PART NUMBER	DESCRIPTION
EDM-760	760000-()	Temperature Indicator

The technical data submitted with your application has been reviewed and accepted as fulfilling the requirements for a TSO authorization and will be retained in our files. For your information, the conditions and tests required for TSO approval are minimum performance standards. The article(s) may be installed on or within a specific type or class of aircraft only if further evaluation by the user/installer documents an acceptable installation that is approved by the Administrator.

The quality control procedures contained in your quality control manual, currently on file at the Los Angeles Manufacturing Inspection District Office, and your statement that those procedures will be applied to the manufacture of the subject article(s) at the above address, are considered adequate in accordance with FAR 21.143.

At completion of the review, J. P. Instruments submitted the following documents for acceptance:

Document Number	Description	Date	Revision
S760-04	Software Configuration Management Plan	07/03/96	1.01
S760-14	Software Verification Results	07/03/96	1.01
S760-16	Software Configuration Index	07/03/96	1.01

Upon the receipt of the aforementioned documents, the FAA has determined that the software system in place is acceptable for J. P. Instruments Model No. EDM-760.

Sincerely,

Rejean Chartier
Rejean Chartier
Manager, Propulsion Branch

United States of America

Department of Transportation—Federal Aviation Administration

Supplemental Type Certificate

Number SA00729SE

This certificate, issued to J. P. Instruments
P.O. Box 7033
Huntington Beach, CA 92646

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified herein meets the airworthiness requirements of Part 21 of the "Regulations."

Original Product—Type Certificate Number: "See attached FAA Approved Model List (AML)
Make: No. SA00729SE for a list of approved airplane
Model: models and applicable airworthiness regulations.

Description of the Type Design Change: Installation of twin temperature indicating system with fuel flow in accordance with J.P. Instruments (JPI) Installation manual Report No. 760, Revision -, dated 7/20/99, or later FAA approved revision.

Limitations and Conditions: Approval of this change in type design applies to the aircraft models listed on the AML only. This approval should not be extended to other aircraft of these models on which other previously approved modifications are incorporated unless it is determined that the relationship between this change and any of those other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that aircraft. A copy of this certificate, Airplane Flight Manual Supplement No. 760-1 dated August 31, 1999, or later FAA approved revision, and FAA Approved Model List (AML) No. SA00729SE must be maintained as part of the permanent records for the modified aircraft.

Cylinder head, oil, turbine inlet and/or exhaust gas temperature, fuel flow equipment, tachometer instruments, and manifold pressure instruments required by the original type design, or if required by other FAA approval, must remain installed.

If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: June 18, 1999

Date of issuance: August 31, 1999

Date revised:

Date amended:



Adrian P. Johnson
Acting Manager, Seattle Aircraft
Certification Office

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

Subject: Permission to use STC.

To Whom It May Concern:

J.P. Instruments holder of STC SA00729SE grants to the purchaser of the EDM-760 series (PN 760000) permission to use STC SA00729SE.

Signed

United States of America
Department of Transportation—Federal Aviation Administration
Supplemental Type Certificate

Number SA4005NM

This certificate, issued to

Precise Flight, Inc.
63354 Powell Butte Road
Bend, OR 97701

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part * of the * Regulations.

Original Product—Type Certificate Number:

Make:

Model:

*See attached FAA Approved Model List (AML)
No. SA4005NM for list of approved airplane models
and applicable airworthiness regulations

Description of the Type Design Change: Precise Flight Inc. Pulselite Control System manufactured in accordance with Precise Flight Engineering Drawing Lists 002P0000, Revision H, dated July 11, 2001; 010P0000, Revision F, dated March 13, 2002; 015P0000, Revision J, dated April 19, 2002, 020P0000; Revision M, dated September 6, 2002; 025P0000, Revision E, dated July 10, 2001, or later FAA approved revisions. Installed in accordance with the appropriate Precise Flight drawing, listed on Precise Flight Engineering Drawing List 001P0020, Revision A, dated July 10, 2002, and/or the Installation Manual listed on FAA AML No. SA4005NM, dated November 27, 2002, or later FAA approved revisions. Maintained in accordance with the appropriate Precise Flight Instructions for Continued Airworthiness found on Engineering Drawing List 001P0020, Revision A, dated July 10, 2002, or found in the appropriate Installation Manual listed on FAA AML SA4005NM, dated November 27, 2002, or later FAA approved revision.

NOTE: Approval for the physical installation of the Pulselite Control Unit will be required if there is no specific reference provided in Engineering Drawing List 001P0020, Revision A, dated July 10, 2002, or later FAA approved revision.

Limitations and Conditions: Approval of this change in type design applies to the model aircraft on the FAA approved AML SA4005NM only. This approval should not be extended to other aircraft of these models on which other previously approved modifications are incorporated unless it is determined by the installer that the relationship between this change and any of those other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that aircraft. A copy of this Certificate, FAA approved AML No. SA4005NM, dated November 27, 2002, or later FAA approved revision, and the Airplane Flight Manual Supplement called out in FAA approved AML SA4005NM, or later FAA approved revision, must be maintained as part of the permanent records for the modified aircraft.

If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: July 21, 1984
Date of issuance: August 19, 1988

Date reissued:
Date amended: 3/31/88; 5/4/89; 6/24/93; 6/22/00;
6/4/01; 11/27/02; 11/7/03



By direction of the Administrator

[Signature]

(Signature)

Acting Manager, Seattle Aircraft Certification Office
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAA FORM 8110-2(10-88)

This certificate may be transferred in accordance with FAR 21.47.

KOSOLA AND ASSOCIATES, INC.
PIPER PA-34 SERIES ONE-PIECE WINDSHIELD KIT
PACKING LIST

PART NAME	PART NUMBER	QTY	PACKED	P.O. NO. / MFG. DATE
WINDSHIELD	95034	1		
() -001 CLEAR	—	—	—	—
() -003 GREEN	—	—	—	—
() -005 GRAY	—	—	—	—
SUPPORT-WELD ASSY	95033-001	1	1	NOV 04 2004
COVER, POST-LOWER	K2036-3	1	1	NOV 04 2004
COVER, POST-LOWER	K2036-4	1	1	24 JUL 2006
BOLT	AN3-10A	11	11	22764
WASHER	AN960-10	11	11	22695
NUT	MS21044N3	11	11	22695
CHERRYLOCK RIVET	CR2163-4-4	8	8	22757
BLOCK	K2036-5	1	1	NOV 04 2004
SCREW	MS35215-55	1	1	20926
SCREW	MS35215-30	2	2	22877
NUT	MS35650-365	2	2	22848
NUT	MS35650-305	1	1	20926
WASHER	NAS1149BN632H	4	4	22848
WASHER	NAS 1149BO332H	2	2	22880
CLAMP	NAS1713D12T	1	1	22804
FAST SET SEALANT	BOSTICK 1100	2	2	23105
DOCUMENT	STC (COPY)	1	1	N/A
INSTALLATION DWG.	97016	1 SET	1	REV. 1



AERO PLASTICS INC.

Heavy Gauge Side Window Installation

Report No. PA28/CON

Installation Drawing List

Applicable Aircraft Models

Piper PA-28-140	S/N 28-20002 through 28-7725290
PA-28-150, -160, -180	S/N 28-1 through 28-4377
PA-28-151	S/N 28-7415001 through 28-7715314
PA-28-161	S/N 28-7716001 through 28-8616057
PA-28-161	S/N 2816001 and up
PA-28-161	S/N 2841001 and up
PA-28-180	S/N 28-4378 through 28-7505261
PA-28R-180	S/N 28R-30005 through 28R-7130019
PA-28-181	S/N 28-7690001 through 28-869056
PA-28-181	S/N 2890001 and up
PA-28R-200	S/N 28R-35001 through 28R-7635545
PA-28-201T	S/N 28-7921001 through 28-7921095
PA-28R-201	S/N 28R-7737002 through 28R-7837317
PA-28R-201	S/N 2837001 through 2837061
PA-28R-201T	S/N 28R-7703001 through 28R-7803372
PA-28R-201T	S/N 2803001 through 2803015
PA-28RT-201	S/N 28R-7918001 through 28R-8218039
PA-28RT-201T	S/N 28R-7931001 through 28R-8631004
PA-28RT-201T	S/N 2831001 through 2831038
PA-28-235	S/N 28-10003 through 28-7710089
PA-28-236	S/N 28-7911001 through 28-861108
PA-28-236	S/N 2811001 and up
PA-44-180	S/N 44-7995001 through 44-8195026
PA-44-180	S/N 4495001 through 4495013
PA-44-180T	S/N 44-8107001 thru 44-8207020

Page Numbers 1 thru 8 Inclusive Rev. 1 12/6/96

086 Boquet Road • Jeannette, PA 15644-4707 • Phone: (724)744-4448 • Fax: (724)744-7372

United States Of America
Department of Transportation - Federal Aviation Administration
Supplemental Type Certificate

Number SA01933LA

This Certificate issued to Garmin AT, Inc.
2345 Turner Road S.E.
Salem, Oregon 97302

*certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part * of the Regulations*

Original Product Type Certificate Number:

* See attached Approved Model List (AML)

Make:

No. SA01933LA for list of approved aircraft

Model:

Models and applicable airworthiness regulations.

Description of Type Design

Change: Installation of Garmin Model 400W / 500W Series GPS-WAAS Navigation System in accordance with FAA Approved Garmin 400W Series Master Data List, Drawing No.: 005-C0221-00, Revision "B", dated October 1, 2007, or later FAA approved revision; or FAA Approved Garmin 500W Series Master Data List, Drawing No.: 005-C0221-01, Revision "B", dated October 1, 2007, or later FAA approved revision. Use applicable FAA approved 400W Series Airplane Flight Manual Supplement, document No. 190-00356-03, Rev. "Original", dated November 20, 2007 or later FAA approved revision; FAA Approved 500W Series Airplane Flight Manual Supplement document No. 190-00357-03, Rev. "Original", dated November 20, 2007 or later FAA approved revision; or FAA Approved Airplane Flight Manual Supplement as defined in Master Data List 005-C0221-00 or 005-C0221-01 defined above.

Limitations and Conditions: This approval should not be incorporated in any aircraft unless it is determined that the interrelationship between this installation and any previous approved configuration will not introduce any adverse effect upon the airworthiness of the aircraft. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator

Date of application: January 31, 2006

Date reissued:

Date of issuance: November 6, 2006

Date amended: November 20, 2007



By direction of the Administrator

(Signature)
Project Manager, Systems & Equipment Branch,
Los Angeles Aircraft Certification Office

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAA Form 8110-2(10-68)

Page 1 of 2

This certificate may be transferred in accordance with FAR 21.47.

United States of America
Department of Transportation—Federal Aviation Administration
Supplemental Type Certificate

Number SA01487SE

This certificate, issued to: Garmin AT, Inc.
2345 Turner Road SE
Salem, OR 97302

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 23 of the Federal Aviation Regulations.

Original Product—Type Certificate Number: *See attached Approved Model List (AML)
Make: No. SA01487SE for list of approved aircraft
Model: models and applicable airworthiness regulations.

Description of the Type Design Change: Installation of a GDL 69 or GDL 69A XM Satellite Receiver in accordance with Garmin Supplemental Type Certificate Master Data List, P/N 005-C0217-00, Revision A, dated December 20, 2004, or later FAA approved revision.

Limitations and Conditions: Compatibility of this design change with previously approved modifications must be determined by the installer. A copy of this certificate must be maintained as part of the permanent records for the modified aircraft.

If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: March 22, 2004

Date reissued:

Date of issuance: December 20, 2004

Date amended:



By direction of the Administrator

Kenneth Bauhant
(Signature)

Acting Manager, Seattle Aircraft Certification Office
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N4542T

Nationality and Registration Mark

Date

1. REMOVED AND REPLACED POINTER ELT, AND ANTENNA. INSTALLED NEW ARTEX ME406 ELT TRANSMITTER AND 110-338 ANTENNA IN SAME LOCATION WHERE OLD ELT WAS REMOVED.
2. THE INSTALLATION WAS DONE IN ACCORDANCE WITH ARTEX INSTALLATION MANUAL 570-1600 REV. C. THE INSTALLATION WAS ALSO DONE USING AC45.13-1B CHAPTER 7 PARAGRAPHS 7-14, 7-16, 7-17, 7-63, 7-64, 7-85, CHAPTER 10, PARAGRAPHS 10-1, 10-2, 10-16, 10-19, 10-20, CHAPTER 11 PARAGRAPHS 11-1, 11-2, 11-8, 11-32, 11-47, 11-48, 11-50, 11-89, 11-186, CHAPTER 12, PARAGRAPHS 12-21, 12-22, AND AC43.13-2B CHAPTERS 2 AND 3 AS A GUIDE.
3. AIRCRAFT WEIGHT AND BALANCE AND EQUIPMENT LIST REVISED.
4. THE ELT TRANSMITTER IS A STAND ALONE UNIT AND IS NOT INTERFACED TO ANY OTHER SYSTEM.
5. FOR CONTINUED AIRWORTHINESS REFER TO THE DESCRIPTION, OPERATION, INSTALLATION AND MAINTENANCE MANUAL DOCUMENT NUMBER: 570-1600 REV. C.
6. _____END_____

☐ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N4542T

06/05/2008

Nationality and Registration Mark

Date

1. REMOVED OLD INSTRUMENT PANEL AND USED AS TEMPLATE. INSTALLED NEW INSTRUMENT PANEL MADE BY JAVELINA CORPORATION FROM MATERIAL .100 QQ-A-250/4 METAL. JAVELINA CORPORATION DRAWING NUMBER 083006-501 WAS USED.
2. REMOVED OLD POST LIGHTS AND REPLACED WITH NEW UMA LIGHT WEDGES AND 10-700-14 INVERTER.
3. THE INSTRUMENT LIGHTS ARE POWERED BY THE MAIN BUSS VIA A 10 AMP CIRCUIT BREAKER LABELED LIGHTS
4. REMOVED OLD DIMMING RHEOSTAT, AND REPLACED WITH NEW AL697-1 LIGHT DIMMING KIT.
5. THE INSTALLATION WAS DONE USING FAA ADVISORY CIRCULAR 43.13-1B PARAGRAPHS 4-1, 4-4, 4-52, 4-53, 4-55, 4-57 7-1, 7-2, 7-14, 7-15, 7-16, 7-17, 7-63, 7-64, 7-85, 7-86, 7-87, 10-1, 10-2, 10-16, 11-1, 11-2, 11-31, 11-32, 11-185, 11-186, 11-187, 11-188, 12-1, 12-2, 12-8, 12-9, 12-10, 12-11, 12-12, 12-14, 12-15, 12-17, 12-18, 12-19, 12-20, 12-21, 12-22, 12-37, AND 12-38, AS A GUIDE. ALSO AC43.13-2B CHAPTERS 1, AND 2 AS A GUIDE.
6. THE OPERATIONAL TEST ON ALL THE AVIONICS SYSTEMS CHECKED GOOD.
7. COMPASS CHECKED NEW COMPASS CARD INSTALLED.
8. WEIGHT AND BALANCE REVISED
9. FOR CONTINUED AIRWORTHINESS USE NORMAL AIRCRAFT MAINTENANCE REQUIREMENTS.
10. _____ END _____

☐ Additional Sheets Are Attached

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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N4542T

06/05/2008

Nationality and Registration Mark

Date

1. INSTALLED NEW TAS MODEL 9900BX Processor, p/n 70-2420, With mounting tray. Transponder coupler, p/n 70-2040 and two L-band antennas.
2. The Display Unit is a GNS 530W and GNS 430W installed in the instrument panel at F.S. 61.5. The GNS 530W and GNS 430W are interfaced to the AVIDYNE/Ryan TAS 9900BX processor for traffic display.
3. The processor was installed in the Mounting Tray Assembly, located at F.S. 220 in rear of aircraft according to instructions in the AVIDYNE/Ryan 9900BX Installation Manual, p/n 32-2351, REV. 04 dated 10/03/2005.
4. The transponder coupler was installed under the floor board at F.S. 120.
5. AN L-band antenna was installed at the top of the aircraft fuselage at F.S. 99 doubler supplied by manufacturer was used.
6. An L- band antenna was installed on the bottom of the aircraft at F.S. 130 doubler supplied by manufacturer was used.
7. The installation was done using; FAA Advisory Circular 43.13-1B, Paragraphs 4-55, 4-57, 7-15, 7-16, 7-17, 7-63, 7-64, 7-85, 7-86, 7-87, 10-1, 10-2, 10-16, 10-19, 10-20, 11-1, 11-2, 11-37, 11-47, 11-48, 11-50, 11-53, 11-56, 11-76, 11-77, 11-85, 11-86, 11-87, 11-96, 11-104, 11-105, 11-106, 11-107, 11-108, 11-117, 11-118, 11-120, 11-121, 11-146, 11-155, 11-158, 11-167, 11-185, 11-186, 11-187, 11-188, 11-230, 11-231, 11-232, 11-235, 12-8, 12-9, 12-10, 12-20, 12-37, as a guide.
8. An electrical load analysis was performed and found that the continuous load of the alternator does not exceed 80% of capacity. The TAS 600 is powered by a 5 amp circuit breaker labeled TAS 600.
9. A complete operational test was performed according to AVIDYNE/Ryan TAS 9900BX Installation Manual, p/n 32-2351, rev. 04 dated October 3, 2005. The equipment Performed satisfactorily and did not adversely affect existing components or systems in the aircraft as required by FAR 23.1301.
10. The aircraft equipment list was revised to reflect these changes; weight and balance data was revised and placed in the aircraft records. An AVIDYNE/Ryan TAS 600 FAA Approved Flight manual supplement dated , _____ was placed in the aircraft.
11. For instructions for continued airworthiness. Refer to Avidyne/Ryan 9900BX INSTALLATION MANUAL p/n 32-2351, rev. 04, dated October 3, 2005, section 1.10, page 8.

12

end

☐ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N4542T

06/05/2008

Nationality and Registration Mark

Date

1. REMOVED OLD HSI AND P/N IN-831A AND INSTALLED NEW SANDEL AVIONICS, INC., SN3500 NAVIGATION DISPLAY P/N SN3500-000.
2. THE SANDEL SN3500 DISPLAY IS INTERFACED TO THE GARMIN GNS 530W FOR NAVIGATION DISPLAY, AND TO THE PMA 8000B MARKER BEACON, AND TO THE KG 102A REMOTE DIRECTIONAL GYRO AND KMT 112 MAGNETIC AZIMUTH TRANSMITTER, AND THE S-TEC SYSTEM 55X AUTO PILOT.
3. INTERFERENCE AND FUNCTIONAL TESTS AND INSPECTIONS WERE ACCOMPLISHED WITH REFERENCE TO ADVISORY CIRCULAR 23.1311.
4. A SYSTEM DESIGN AND ANALYSIS WAS CONDUCTED WITH REFERENCE TO ADVISORY CIRCULAR 23.1309-1.
5. FEDERAL AVIATION REGULATIONS, 23.1301, 23.1309(A,B,D), 23.1311, 23.1321(A,B,D), 23.1322, 23.1327(A), 23.1331, 23.1351, 23.1357(A-D), 23.1365, 23.1381, 23.1529, AND 23.1581 WERE THE BASIS OF COMPLIANCE.
6. THE INSTALLATION WAS DONE USING SANDEL SN3500 PRIMARY NAVIGATION DISPLAY INSTALLATION MANUAL REVISION D, DATED 11/15/07 AS A GUIDE.
7. THE INSTALLATION WAS ALSO DONE USING FAA ADVISORY CIRCULAR 43.13-1B PARAGRAPHS 4-1, 4-4, 4-52, 4-53, 4-55, 4-57 7-1, 7-2, 7-14, 7-15, 7-16, 7-17, 7-63, 7-64, 7-85, 7-86, 7-87, 10-1, 10-2, 10-16, 11-1, 11-2, 11-31, 11-32, 11-185, 11-186, 11-187, 11-188, 12-1, 12-2, 12-8, 12-9, 12-11, 12-14, 12-15, 12-17, 12-18, 12-37, AND 12-38, AS A GUIDE. ALSO AC43.13-2B CHAPTERS 1, AND 2 AS A GUIDE.
8. INSTALLATION APPROVAL IS SOUGHT WITH REFERENCE TO FLIGHT STANDARDS INFORMATION BULLETIN, FSAW 95-09, TITLED "ELECTRONIC HORIZONTALSITUATION INDICATOR (EHSI) APPROVALS".
9. COMPASS CHECKED NEW COMPASS CARD INSTALLED.
10. WEIGHT AND BALANCE AND EQUIPMENT LIST REVISED.
11. FOR CONTINUED AIRWORTHINESS SEE SANDEL SN3500 PRIMARY NAVIGATION DISPLAY INSTALLATION MANUAL REVISION D, DATED 11/15/07.
12. _____ END _____

☐ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N4542T

06/05/2008

Nationality and Registration Mark

Date

1. REMOVED OLD AVIONIC EQUIPMENT CONSISTING OF: KLN 90B GPS, KX 155 COM/NAV, KN 62 DME, ALPINE AM/FM STEREO AND CD CHANGER, GEM 602 LEFT AND RIGHT ENGINE MONITORS, AT50A TRANSPONDER, KI 208 INDICATOR, GLS 350 GLIDE SLOPE, KMA 24 AUDIO PANEL, KR 85 ADF RECEIVER, KI 225 ADF INDICATOR, CA814A AUTO-PILOT COMPUTER, PS 815B POWER SUPPLY, VHF 251 COM, VIR 351 NAV, AND A R.C. ALLEN TURN AND SLIP INDICATOR.
2. INSTALLED NEW GARMIN GTX 327 TRANSPONDER IN SAME PLACE THE AT 50A TRANSPONDER WAS REMOVED IN PILOT'S INSTRUMENT PANEL AT F.S. 61.
3. THE INSTALLATION WAS DONE IN ACCORDANCE WITH THE MANUFACTURER'S INSTALLATION MANUAL P/N 190-00207-02 REVISION 1, DATED MAY 2002. AND HARDWARE SUPPLIED BY THE MANUFACTURER. THE EXISTING ANTENNA'S WAS USED.
4. INSTALLED NEW PS ENGINEERING PMA 8000B AUDIO PANEL IN SAME LOCATION WHERE KMA 24 AUDIO PANEL WAS REMOVED.
5. INSTALLED KG 102A REMOTE GYRO IN REAR OF AIRCRAFT AT FS 156. THE KG 102A IS INTERFACED TO THE SANDELL SN3500 NAVIGATION DISPLAY. REMOVED AND REPLACED FLUX DETECTOR WITH KMT 112 FLUX DETECTOR.
6. THE INSTALLATION WAS ALSO DONE USING FAA ADVISORY CIRCULAR 43.13-1B PARAGRAPHS 4-1, 4-4, 4-52, 4-53, 4-55, 4-57 7-1, 7-2, 7-14, 7-15, 7-16, 7-17, 7-63, 7-64, 7-85, 7-86, 7-87, 10-1, 10-2, 10-16, 11-1, 11-2, 11-31, 11-32, 11-185, 11-186, 11-187, 11-188, 12-1, 12-2, 12-8, 12-9, 12-11, 12-14, 12-15, 12-17, 12-18, 12-37, AND 12-38, AS A GUIDE. ALSO AC43.13-2B CHAPTERS 1, AND 2 AS A GUIDE.
7. COMPASS CHECKED NEW COMPASS CARD INSTALLED.
8. WEIGHT AND BALANCE AND EQUIPMENT LIST REVISED.
9. FOR CONTINUED AIRWORTHINESS SEE MANUFACTURER'S INSTALL MANUALS AND FAA ADVISORY CIRCULAR 43.13-1B CHAPTER 12.
10. _____

END

☐ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Installed pneumatic door seals in accordance with STC SA4234WE, Bob Fields Aerocessories. Weight and Balance revised. Nothing follows

END

☐ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Installed Brackett airfilter BA-105 Per manufactures installtion
sheet BA-105-3 and STC SA693CE.
-----END-----

☐ Additional Sheets Are Attached

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Rajay STC SA2937WE incorporated RJ-0913-11 and RJ-0913-21 turbo supports for the PA34-200 LH engine and RJ-0913-12 and RJ-0913-22 supports for the RH engine. These supports or struts supported the original type design turbochargers RJ-0329-6, LH and RJ-0329-8, RH.

The original type design turbochagers has been upgrade to a new design which changes the bolt patterns so they will not match the the original struts.

Instruction provided by Rajay to alter the struts ends to incorporate the upgraded turbochargers. New attachment brackets replaced old brackets. Old brackets were cut away and the new ones were installed one the struts. Brackets were welded into place using practices found in AC43.13-1A Section 2 Fig 2.1, 2.2.

No weight change incoperated.

-----END-----

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

- A. INSTALLED 1 EA KING KMA 24 AUDIO PANEL SR# 25338, 1 EA NORTH-STAR M1-A LORAN SR# N16141, 1 EA NORTHERN AIRBORNE TECH RS08-001 RELAY UNIT SR# 3848, 1 EA SIGTRONICS SPA 600 ICS SR# 1100383, 1 EA RES 600 SIGTRONIC STEREO SWITCHING UNIT SR# 1050145, 1 EA AVIONICS WEST EC-200 AM/FM STEREO CASSETTE SR# 05013657.
- B. ALL WORK DONE IN ACCORDANCE WITH AC 43.13-2A PARA. 22.
- C. PLACKARD FOR LORAN INSTALLED IN ACCORDANCE WITH AC 20.-121, PARA.3.
- D. LORAN INSTALLATION COMPLIES WITH AC 90-45A APENDIX A PARA. 2.C. AND 3.C. (1).
- E. WEIGHT AND BALANCE COMPUTED
EQUIPMENT LIST AMENDED
LOG BOOK ENTRIES MADE.

END

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.) N 4542T
October 6, 1980

Installed General Fire Extinguisher, Model GH2½ J Halon 1211 on an .064 aluminum plate clamped to the structure under the copilot seat.

Equipment list and weight and balance revised.

END=====

☐ ADDITIONAL SHEETS ARE ATTACHED

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

6. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

REMOVED: 1. Bonzer TRN-71 Radar Altimeter
2. King KN65 DME
3. Foster Airdata 511 RNAV
4. Collins TDR 950 Transponder
5. Smith Encoding Altimeter
6. IVSI Indicator

INSTALLED: 1. KING KN 62 DME
2. AEROSONIC ENCODING ALTIMETER
3. VERTICAL SPEED INDICATOR
4. HOSKINS CFS 2000 COMPUTERIZED FUEL SYSTEM PER STC SA 154RM

EQUIPMENT LIST AND WEIGHT AND BALANCE REVISED.

*****END*****

Record No: 16279



Ship Shape

Custom Products, Inc.

P.O. Box 1141
Mooresville, NC 28116
(704) 663-4159 Fax 663-7904

Tested For: **PERRONE AEROSPACE**

Inv/PO No: 1664SH

P/N: 1664

P/N 2:

P/N 3:

P/N 4:

MFG P/N:

Color: LIGHT SILVER

Time In Conditioning 5/19/08 1:00PM

Test Date: May 20, 2008

Test Report: 052008

Pattern: SHEARLING

D/L No: 221420

Time Out Conditioning 5/20/08 1:30PM

TWELVE SECOND VERTICAL FLAMMABILITY TEST RESULTS

14 CFR 25.853 (a) Appendix F Part I (a) (1) (II) (Amendment 25-83)

	SELF-EXTINGUISH TIME (seconds)	BURN LENGTH (Inches)	EXTINGUISH TIME (seconds)
	1.80	0.50	0.00
	2.50	0.70	0.00
	3.20	0.50	0.00
Average:	<u>2.50</u>	<u>0.57</u>	<u>0.00</u>
Average:			
Max:	15.0	8.0	5.0

TESTED BY: B. W. How

Passed: ☒ Failed: ☐