



DETAILS CONSULTANT

Structural Calculations

Job No.	DC23-038	Revision No.	00
Building	Skylight	Revision Date	16-SEPT-23
Project Name	Proposed Senior & Junior skylight structural glazing		
Project Address	Western Port, Secondary College, 215 High St., Hastings, VIC 3195		
Client Name	ACOL Skylight		
Client Address	26 Simcock St., Somerville, VIC 3912		

Designed By	Kamal Kouli CPEng-NER-RPEQ
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Total Pages
18

Revision Register

Rev. No.	Revision Date	Description of Revision
00	16-SEPT-23	Original release

Preface

This structural calculation package has been prepared by Details Consultant using the latest applicable Australian Codes and Standards along with the latest developments in engineering practices.

Computer software analysis programs such as Space Gass, ETABS and Limcon have been used where applicable to determine structural requirements. The output of these programs has been incorporated in this package together with explanation text where possible.

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These calculations are valid only for the specific project address stated on the Structural Calculations cover page and must not be used for a different site without the knowledge and written consent from Details Consultant.

Please use this structural calculation in conjunction with the approval drawings provided.

For any questions regarding this package please do not hesitate to contact our office. Details Consultant keeps records of reference on structural calculations for each job.

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	Address:	Western Port, Secondary College, 215 High St.	Job No.:	DC23-038
		Hastings, VIC 3195		

2-Load Estimations

Dead Loads:

From Table 6.1 -AS 2188 :

TABLE 6.1 DEAD LOADS PERPENDICULAR TO THE GLASS DUE TO SELF-WEIGHT				
Angle of glass to the horizontal (degrees)	Single glazing		Insulated glass unit (IGU)	
	Maximum (kPa)	Minimum (kPa)	Maximum (kPa)	Minimum (kPa)
0	0.57	0.13	1.14	0.26
5	0.57	0.13	1.13	0.26
10	0.56	0.13	1.12	0.26

For roof slope 10 degree and for IGU (6+6) panels we can find that :

Maximum weight= 1.12 KPa THus for 12+12 panels = 2*1.12=2.24 KPa

Minimum weight =0.26 KPa THus for 12+12 panels = 2*0.26 =0.52 KPa

Live Loads:

Uniform load is 0.25 Kpa

Point Load =1.1 KN

Wind Loads:

Important factor= 3 probability (1/1000) Vs= 37 m/sec (Vs/Vu)^2= 0.65
Vu*= 46 m/sec

Wind region = A

TC= 2.5

Average building Height= 12 m

Mzcat= 0.95

Md= 1

Ms= 1 +Ns

Mt= 1 Flat

Pu= 0.0006* (Vu x Mzcat x Md x Mt x Ms)^2 x Cfig = 1.146 x Cfig Kpa

Cpe= -0.9 0.4

Cpi= 0.2 -0.3

Cpn= 1.1 0.7

Kc= 0.9 (intenral and external combination)

Kl= 1 Positive pressure

2 Negative pressure,roof edges, ridges, hips

Cfig= Cpn x Kc x Kl (2)= -1.98 edge

Cpn x Kc x Kl (1)= 0.63 general

Pu* (ULS)= -2.27 Kpa Edge Ps* (SLS)= -1.47 Kpa Edge
0.72 Kpa General 0.47 Kpa General

Combinations:

U1= DL max+1.5LL= 2.24+1.5*0.25=+2.615Kpa

U3=DL max+Wind Down=2.24+0.72=2.97Kpa

U4=DLmin-Wind Up=0.52-2.27=-1.75 Kpa

S1=DI+0.7*LL=0.6+0.7*0.25=0.775 Kpa

S2=DL+Ws=0.60+0.47=1.07Kpa

S3=0.9DL-Ws=0.9*0.52-1.47=-1 Kpa

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3-Check allowable span B based on wind pressure applied

Two edge supported 1.25m

Pu= 2.97 Kpa

Maximum span B= 2000 mm
<2000
--> Maximum span 2000 mm

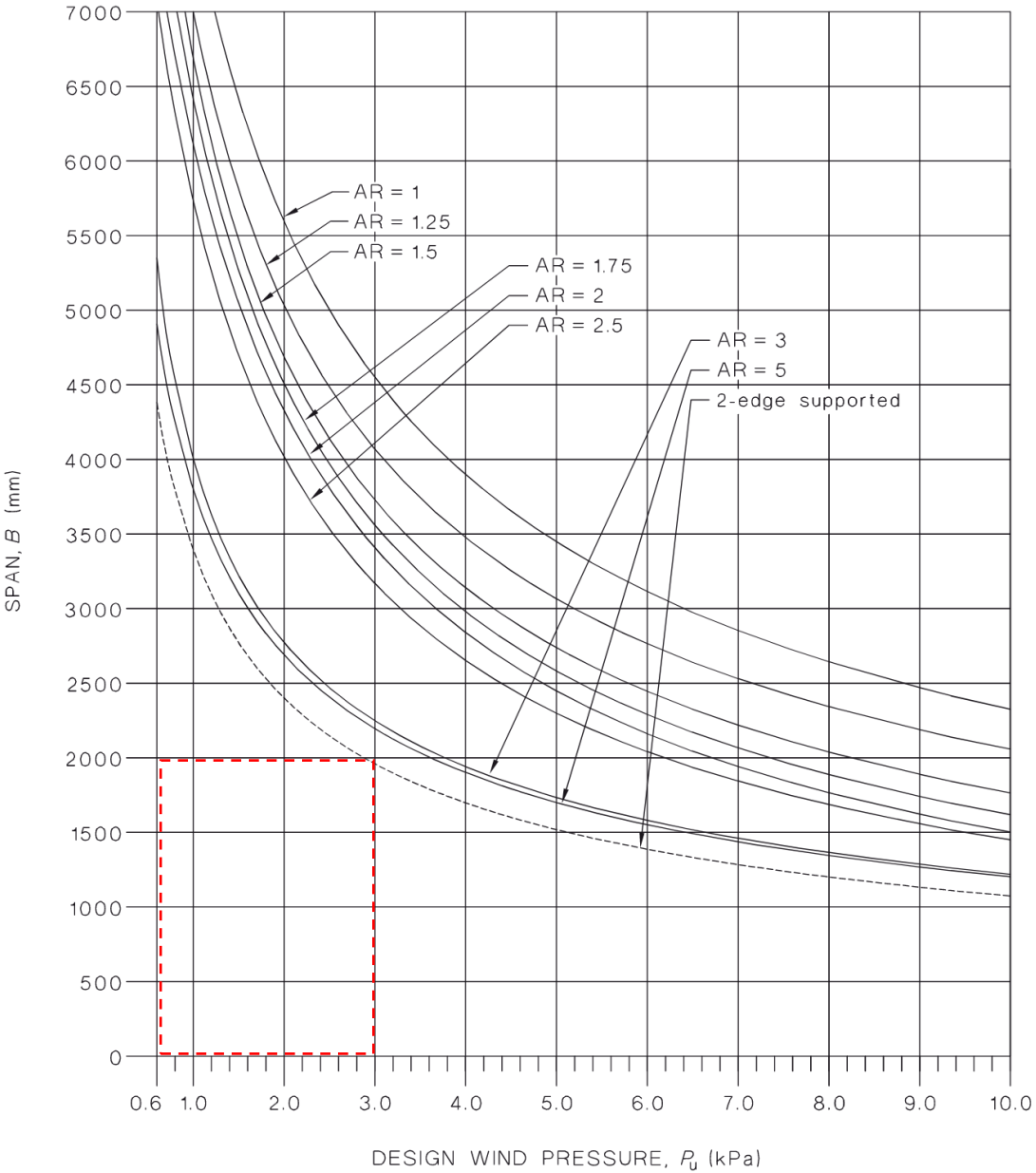


FIGURE 4.16 MAXIMUM SPAN FOR MONOLITHIC 12 mm TOUGHENED GLASS

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4-Check minimum allowable thickness

Two edge supported 1.25m

Ps= 1.07 Kpa

Max B/t= 175

B= 1250 mm

t= 12 mm

B/t= 104.2

104.2 < 175
OK

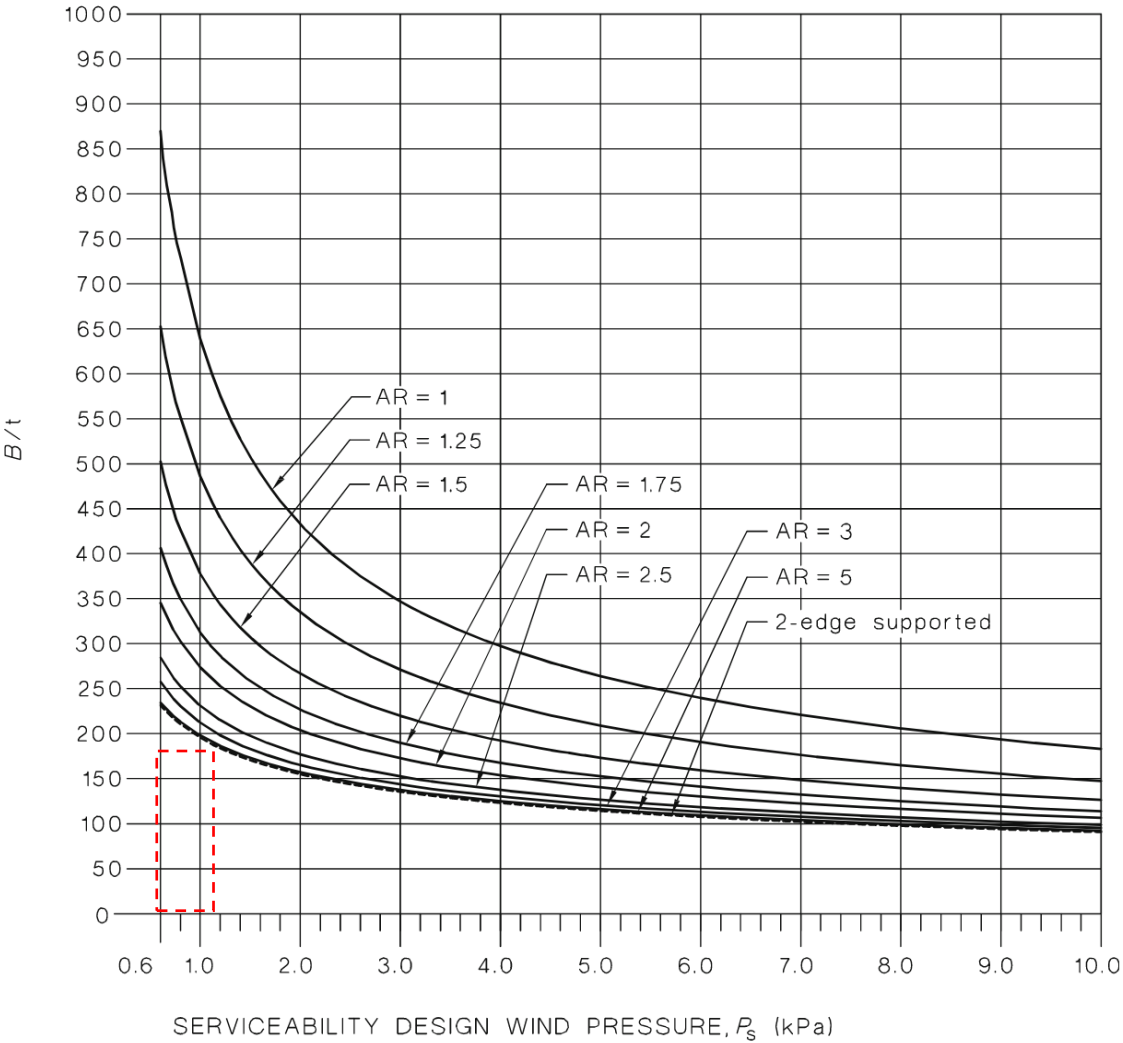



FIGURE 4.35 CURVES FOR B/t ALLOWABLE FOR DEFLECTION OF GLASS LIMITED TO SPAN/60

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Glass panels Calculation

We will model the glass panels with supporting steel framing using Space Gass program. Then we will check the toughened glass tensile stresses on both edge & center. We will check the deflections and compare the results with allowed values per code.


Load cases

1	SW-MAX	1.2*selfweight-DOWN WARDS
2	LL UNIFORM	UNIFORM
3	Wup	winds up
4	Wdown	wind down
5	LL-POINT1	POINT ON EDGE
6	SW-MIN	0.9*selfweight- FOR UPLIFT
7	SW without factor	deflection
8	LL-POINT2	POINT IN CENTER
10	SWmax+1.5LLuniform	
11	SWmax+Wdown	
12	SWmin+Wup	
13	SWmax+1.5LL point1	
14	SW +0.7LLuniform	deflection uniform
15	SW +LLpoint1	deflection point 1
16	SWmax+1.5LL point2	
17	SW +LL point2	deflection point2

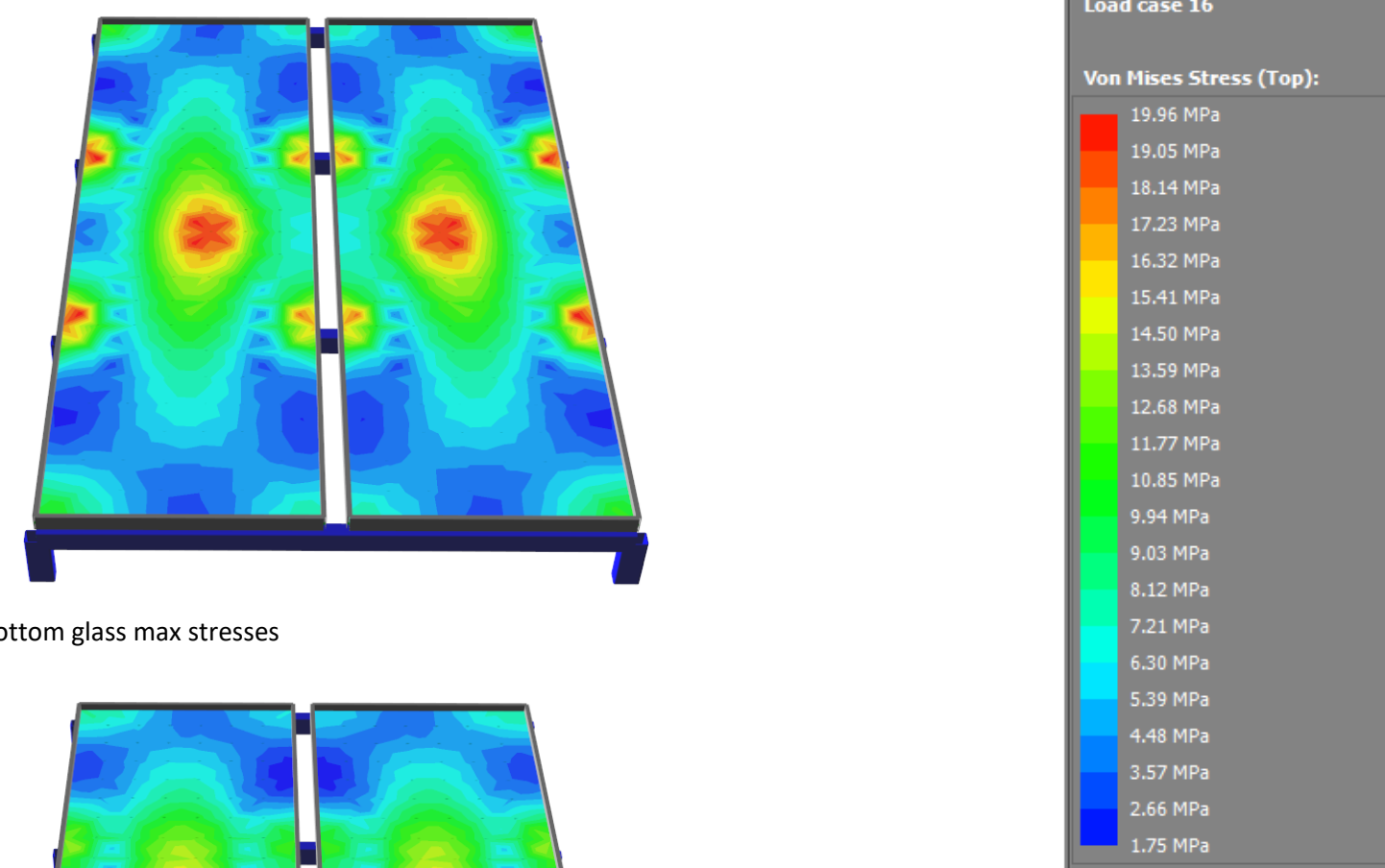
Load combinations

Combination Load Cases										
Combination Case	Title	1 SW-MAX	2 LL UNI...	3 Wup	4 Wdown	5 LL-POI...	6 SW-MIN	7 SW wit...	8 LL-POI...	9 S
*	Click here to add a new row									
10	SWmax+1.5LLuniform	1	1.5							
11	SWmax+Wdown	1			1					
12	SWmin+Wup			1			1			
13	SWmax+1.5LL point1	1				1.5				
14	SWmax+1.5LL point2	1							1.5	
30	SW +0.7LLuniform		0.7					2.4		
31	SW +LLpoint1					1		2.4		
32	SW +LL point2							2.4	1	

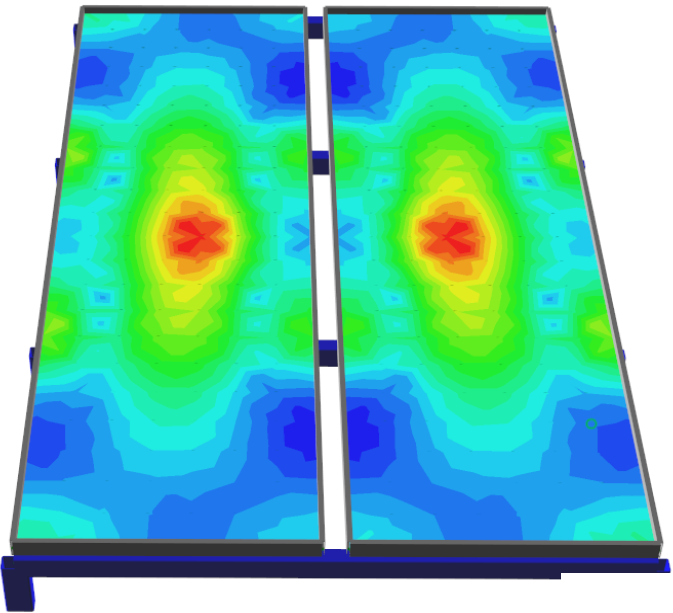
Toughened glass C3=0.5 -->
DLmax=2.4x weight

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1-Top glass max stresses




2-Bottom glass max stresses

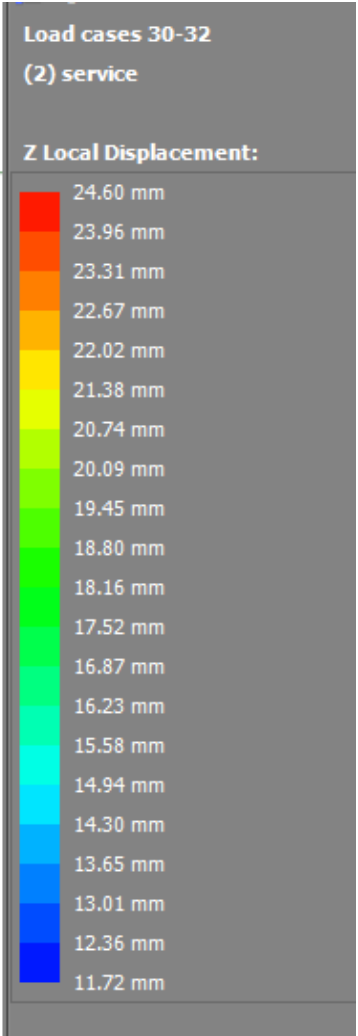
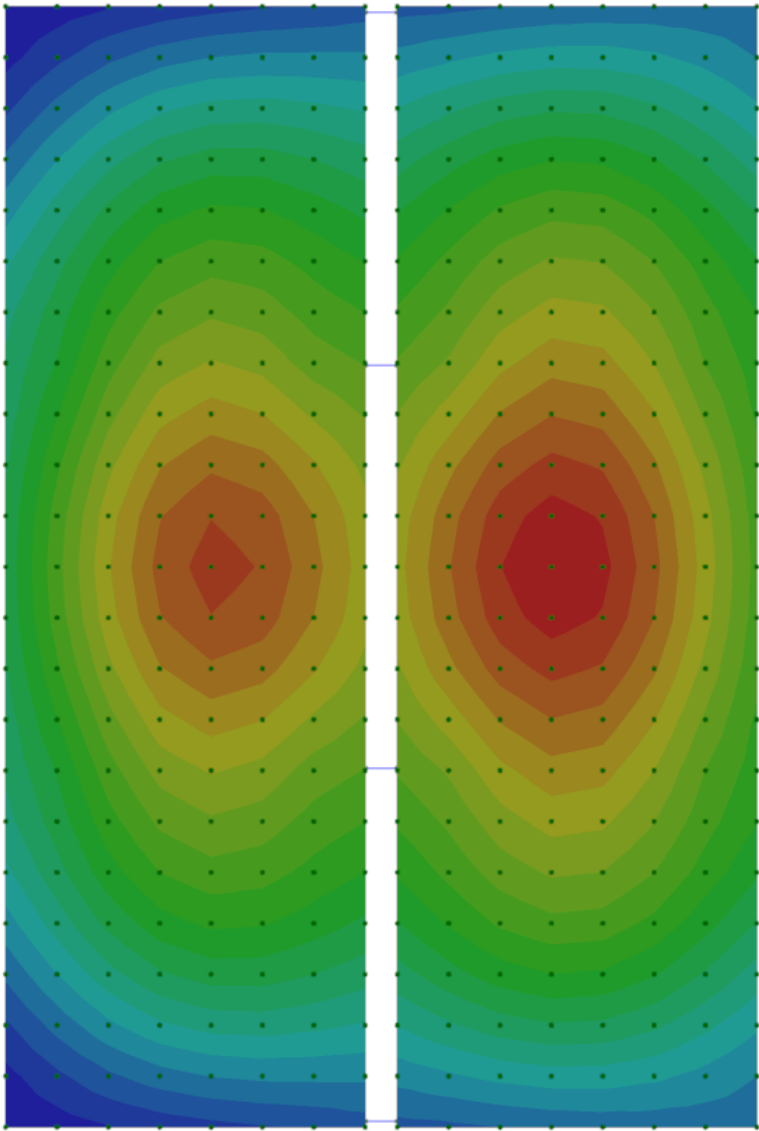


Maximum stresses in glass is 19 Mpa <63.16 Mpa
OK

Glass type	Nominal thickness (mm)	Ultimate limit state design stress at locations shown	
		Away from edge (MPa)	At edge (MPa)
Annealed	3	41.00	32.80
	4	38.99	31.19
	5	37.45	29.96
	6	36.20	28.96
	8	34.33	27.46
	10	32.80	26.24
	12	31.57	25.25
	15	30.15	24.12
	19	28.72	22.98
Toughened	25	26.96	21.57
	4	97.47	77.97
	5	93.61	74.89
	6	90.49	72.39
	8	85.82	68.65
	10	82.01	65.61
	12	78.91	63.13
	15	75.37	60.30
	19	71.81	57.45
	25	67.41	53.93

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3-Deflection:



Allowable deflection $L/60=1250/60=20.83\text{mm}$
We have 12+12 mm IGU panel, each panel share $k_{\text{pane}}=0.625$ of the loads as per 3.4.2

3.4.2 Insulating glass units (IGU)

For insulating glass units, each pane shall be checked for both the ultimate strength and serviceability limit state conditions with the load contribution to the pane determined from k_{pane} , as follows:

$$k_{\text{pane}} = \frac{1.25t_{\text{pane}}^3}{\sum_i t_i^3} \leq 1$$


where

- k_{pane} = load-sharing factor of pane being checked
- t_{pane} = thickness of pane being checked (including laminated glass as per Clause 3.4.1 and glass thickness as per Clause 3.6 or Table 4.1)
- t_i = thickness of each pane of glass within the assembly (see Clause 3.6)
- i = total number of panes within the assembly

NOTE: For insulating glass units with two panes of equal thickness $k_{\text{pane}} = 0.625$.

Deflection of each panel is $=24.60\text{mm} \times 0.625=15.375\text{mm} < 20.83\text{mm}$ OK

OK

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Aluminum frame Calculation

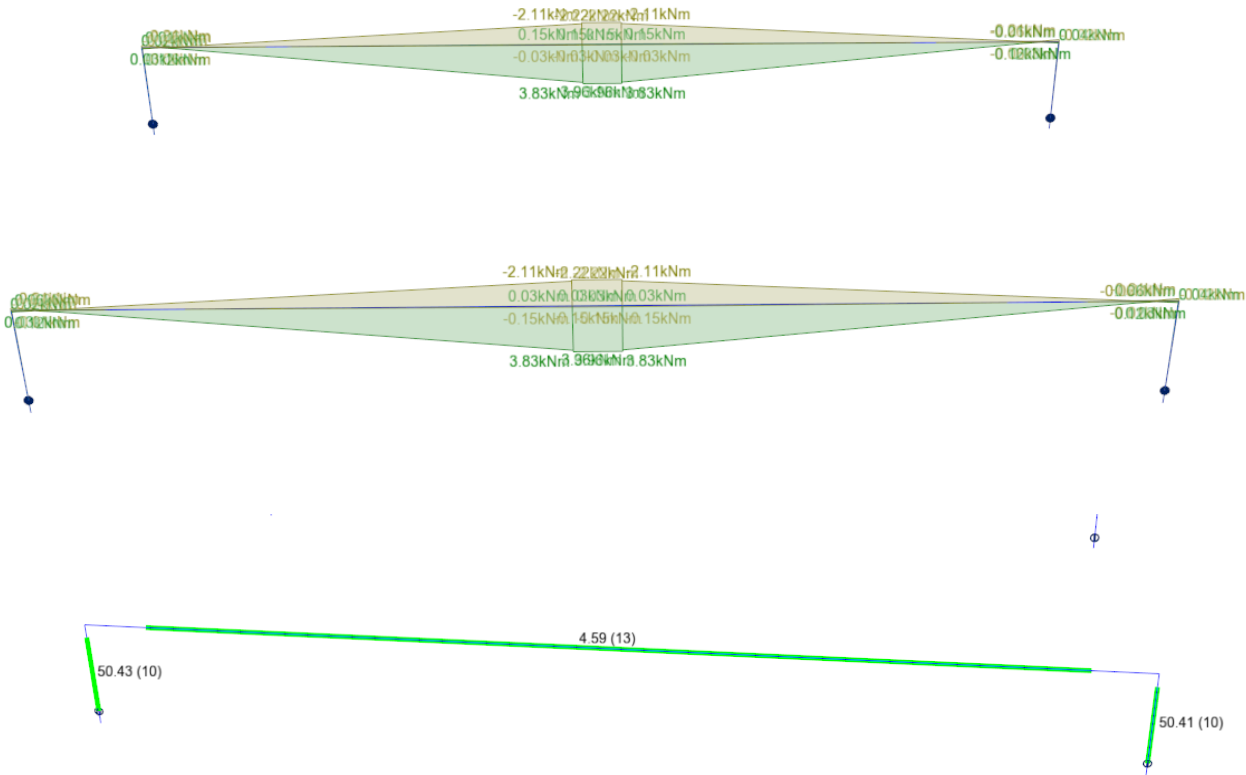
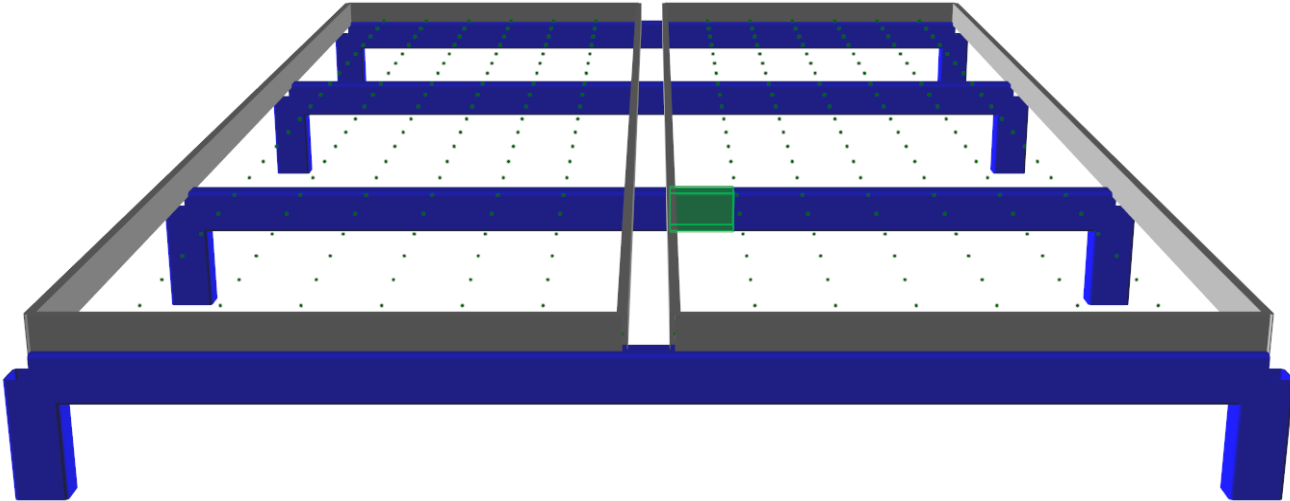
Max stresses in mid Aluminum frame is 0.69 KN.m


Adopt 80x50x3 RHS Aluminum section

OK

Check sheet No17-18

Steel frame Calculation



 DETAILS CONSULTANT	Project:	School hub Skylights (Junior)	Sheet:	10
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2-Load Estimations

Dead Loads:

From Table 6.1 -AS 2188 :

TABLE 6.1
DEAD LOADS PERPENDICULAR TO THE GLASS DUE TO SELF-WEIGHT

Angle of glass to the horizontal (degrees)	Single glazing		Insulated glass unit (IGU)	
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For roof slope 10 degree and for IGU (6+6) panels we can find that :
Maximum weight= 1.12 KPa THus for 12+12 panels = 2*1.12=2.24 KPa
Minimum weight =0.26 KPa THus for 12+12 panels = 2*0.26 =0.52 KPa

Live Loads:

Uniform load is 0.25 Kpa
Point Load =1.1 KN

Wind Loads:


Important factor= 3 probability (1/1000) Vs= 37 m/sec (Vs/Vu)^2= 0.65
Vu*= 46 m/sec

Wind region = A
TC= 2.5
Average building Height= 12 m
Mzcat= 0.95
Md= 1
Ms= 1 +Ns
Mt= 1 Flat

Pu= 0.0006* (Vu x Mzcat x Md x Mt x Ms)^2 x Cfig = 1.146 x Cfig Kpa
Cpe= -0.9 0.4
Cpi= 0.2 -0.3
Cpn= 1.1 0.7
Kc= 0.9 (intenral and external combination)
Kl= 1 Positive pressure
2 Negative pressure,roof edges, ridges, hips
Cfig= Cpn x Kc x Kl (2)= -1.98 edge
Cpn x Kc x Kl (1)= 0.63 general
Pu* (ULS)= -2.27 Kpa Edge Ps* (SLS)= -1.47 Kpa Edge
0.72 Kpa General 0.47 Kpa General

Combinations:
U1= DL max+1.5LL= 2.24+1.5*0.25=+2.615Kpa
U3=DL max+Wind Down=2.24+0.72=2.97Kpa
U4=DLmin-Wind Up=0.52-2.27=-1.75 Kpa

S1=DI+0.7*LL=0.6+0.7*0.25=0.775 Kpa
S2=DL+Ws=0.60+0.47=1.07Kpa
S3=0.9DL-Ws=0.9*0.52-1.47=-1 Kpa

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3-Check allowable span B based on wind pressure applied

Two edge supported 1.25m

Pu= 2.97 Kpa

Maximum span B= 2000 mm
 <2000
 --> Maximum span 2000 mm

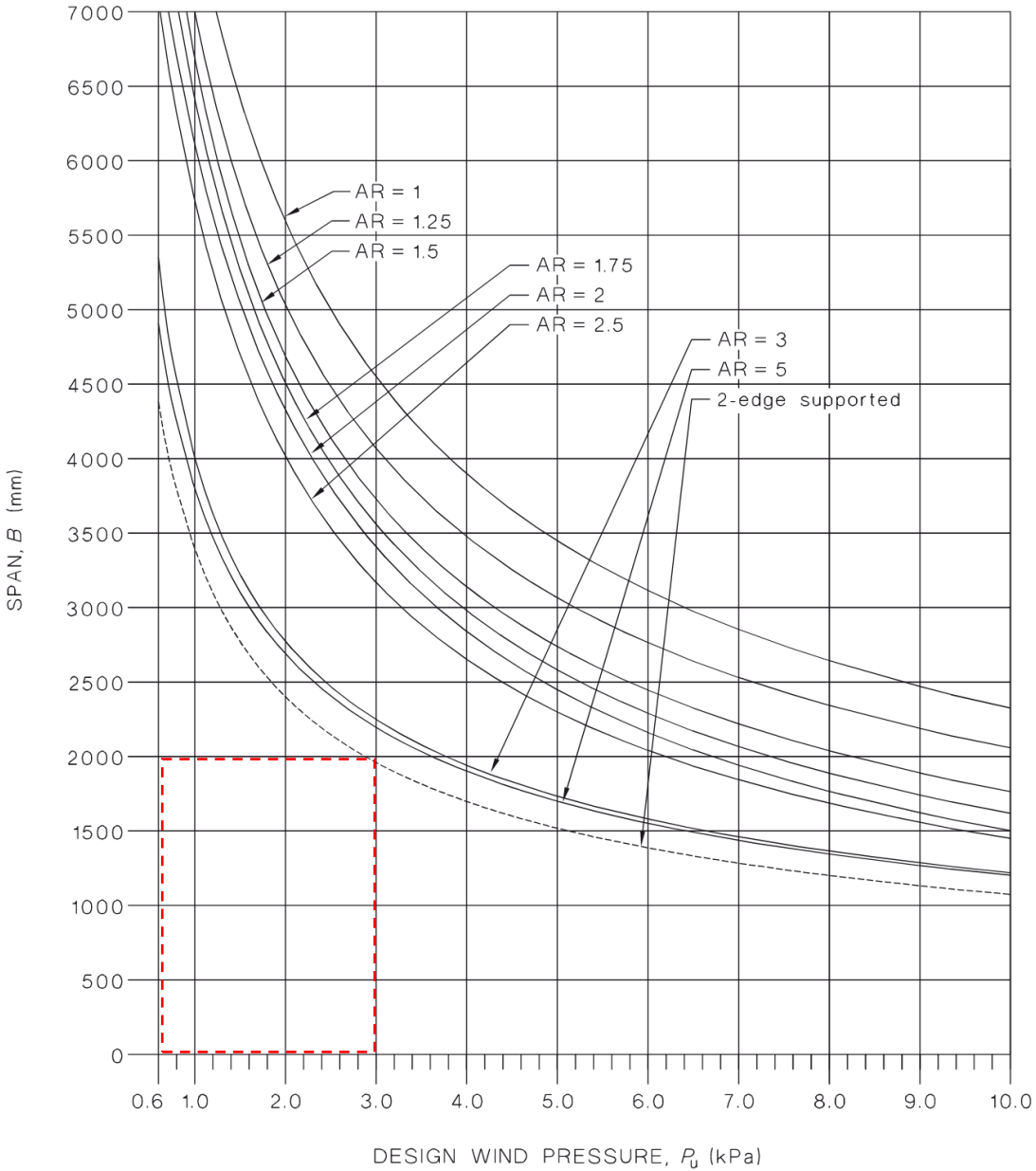



FIGURE 4.16 MAXIMUM SPAN FOR MONOLITHIC 12 mm TOUGHENED GLASS

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4-Check minimum allowable thickness

Two edge supported 1.25m

Ps= 1.07 Kpa

Max B/t= 175

B= 1250 mm

t= 12 mm

B/t= 104.2

104.2 < 175
OK

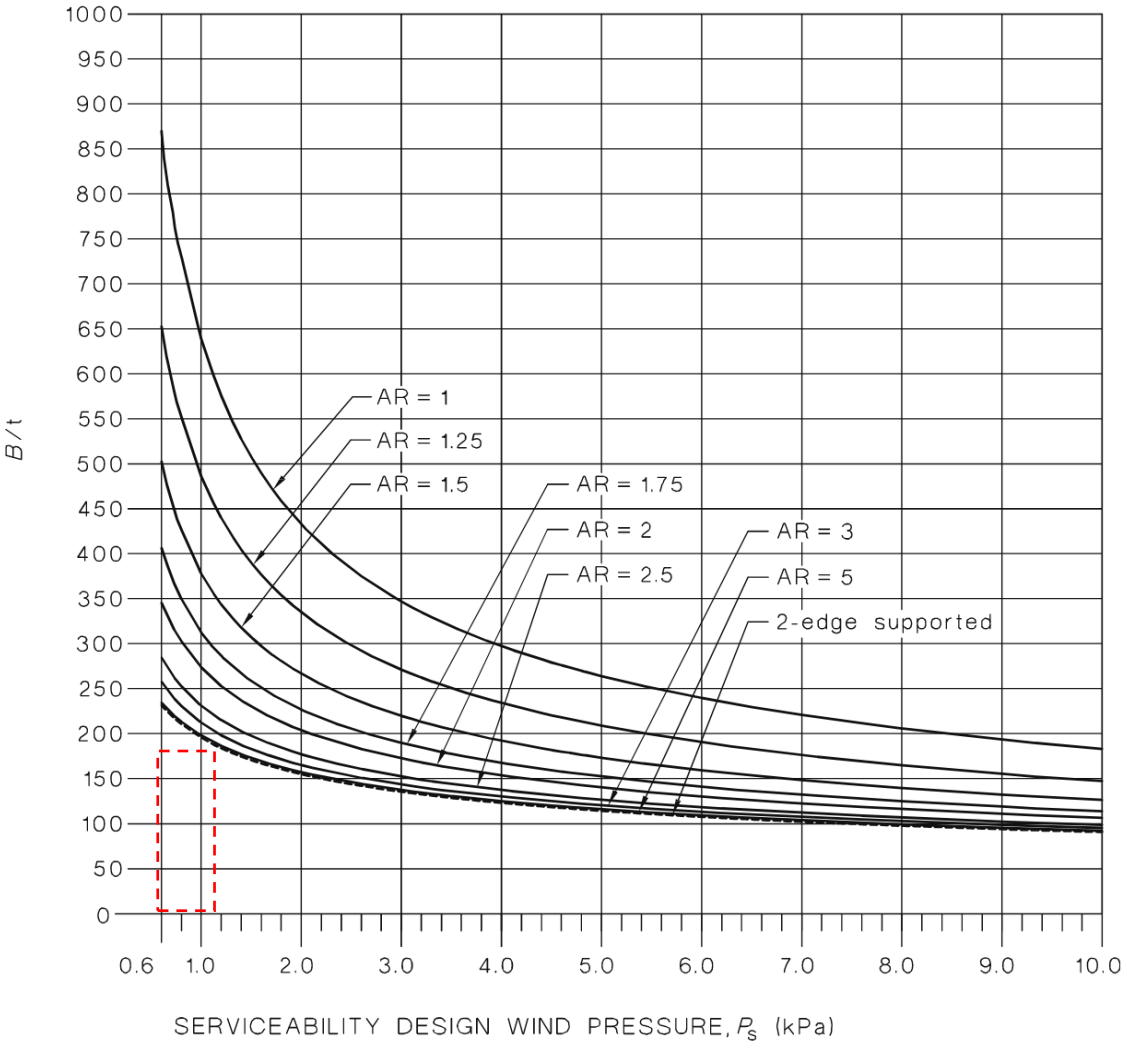



FIGURE 4.35 CURVES FOR B/t ALLOWABLE FOR DEFLECTION OF GLASS LIMITED TO SPAN/60

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Glass panels Calculation

We will model the glass panels with supporting steel framing using Space Gass program. Then we will check the toughened glass tensile stresses on both edge & center. We will check the deflections and compare the results with allowed values per code.

Load cases

1	SW-MAX	1.2*selfweight-DOWN WARDS
2	LL UNIFORM	UNIFORM
3	Wup	winds up
4	Wdown	wind down
5	LL-POINT1	POINT ON EDGE
6	SW-MIN	0.9*selfweight- FOR UPLIFT
7	SW without factor	deflection
8	LL-POINT2	POINT IN CENTER
10	SWmax+1.5LLuniform	
11	SWmax+Wdown	
12	SWmin+Wup	
13	SWmax+1.5LL point1	
14	SW +0.7LLuniform	deflection uniform
15	SW +LLpoint1	deflection point 1
16	SWmax+1.5LL point2	
17	SW +LL point2	deflection point2

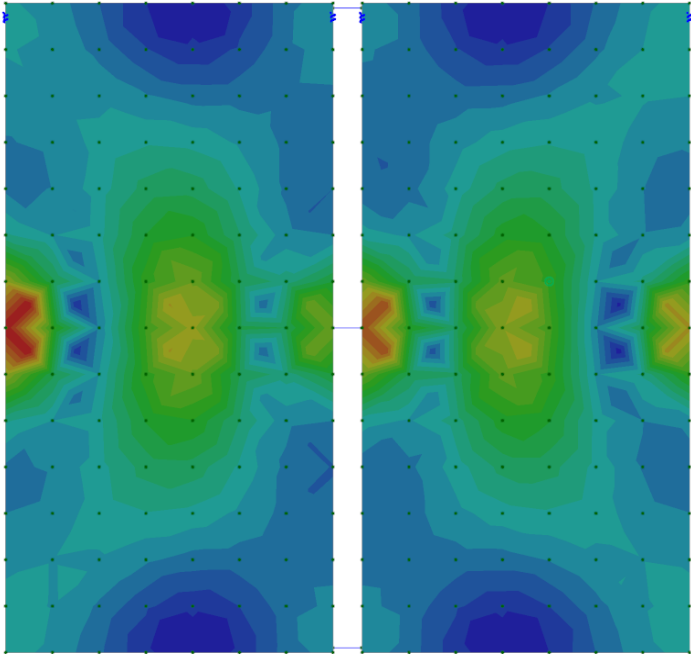
Load combinations

Combination Load Cases										
Combination Case	Title	1 SW-MAX	2 LL UNI...	3 Wup	4 Wdown	5 LL-POI...	6 SW-MIN	7 SW wit...	8 LL-POI...	9 S
*	Click here to add a new row									
10	SWmax+1.5LLuniform	1	1.5							
11	SWmax+Wdown	1			1					
12	SWmin+Wup			1			1			
13	SWmax+1.5LL point1	1				1.5				
14	SWmax+1.5LL point2	1							1.5	
30	SW +0.7LLuniform		0.7					2.4		
31	SW +LLpoint1					1		2.4		
32	SW +LL point2							2.4	1	

Toughened glass C3=0.5 -->
DLmax=2.4x weight

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1-Top glass max stresses



2-Bottom glass max stresses

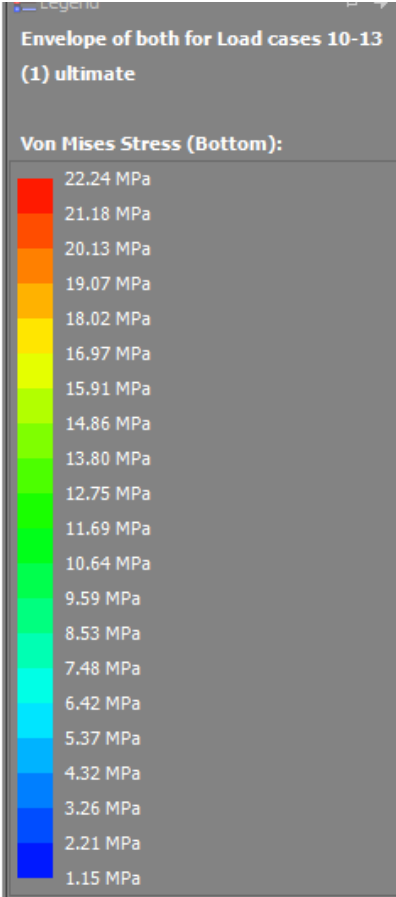
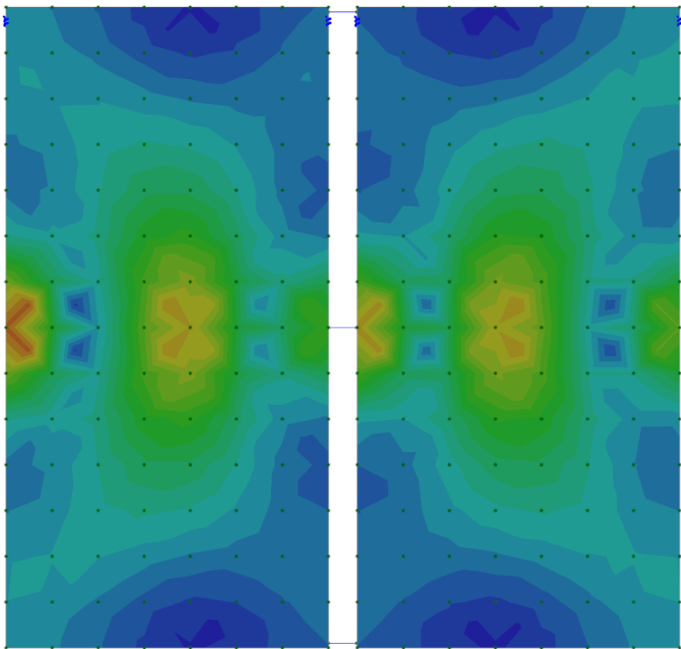



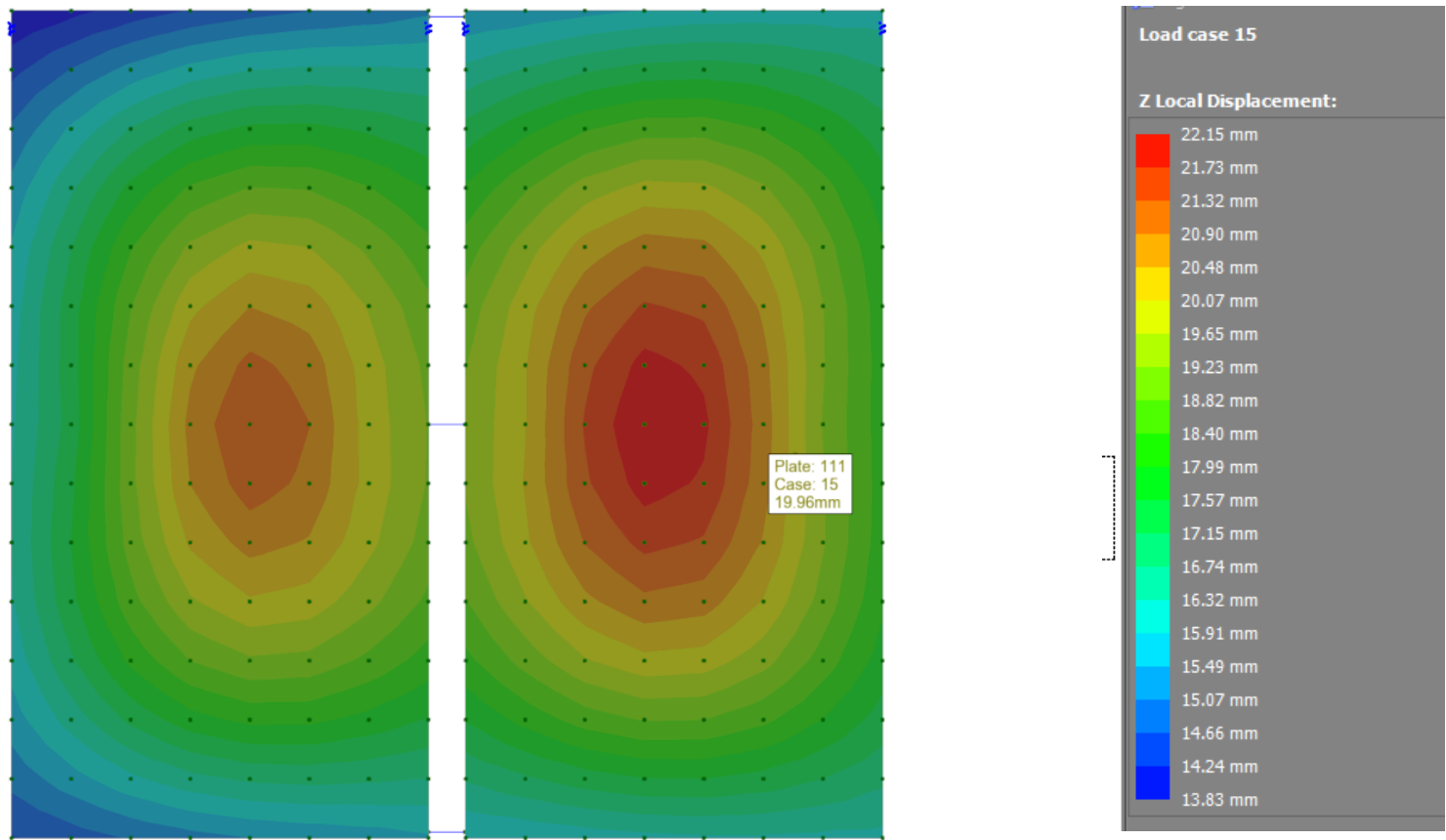
TABLE B1
ULTIMATE LIMIT STATE DESIGN STRESSES FOR GLASS SUBJECTED TO WIND LOADING

Glass type	Nominal thickness (mm)	Ultimate limit state design stress at locations shown	
		Away from edge (MPa)	At edge (MPa)
Annealed	3	41.00	32.80
	4	38.99	31.19
	5	37.45	29.96
	6	36.20	28.96
	8	34.33	27.46
	10	32.80	26.24
	12	31.57	25.25
	15	30.15	24.12
	19	28.72	22.98
Toughened	25	26.96	21.57
	4	97.47	77.97
	5	93.61	74.89
	6	90.49	72.39
	8	85.82	68.65
	10	82.01	65.61
	12	78.91	63.13
	15	75.37	60.30
	19	71.81	57.45
	25	67.41	53.93

Maximum stresses in glass is 22.2 Mpa <63.16 Mpa
OK

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3-Deflection:



Allowable deflection $L/60=1250/60=20.83\text{mm}$
We have 12+12 mm IGU panel, each panel share $k_{\text{pane}}=0.625$ of the loads as per 3.4.2

3.4.2 Insulating glass units (IGU)

For insulating glass units, each pane shall be checked for both the ultimate strength and serviceability limit state conditions with the load contribution to the pane determined from k_{panes} as follows:


$$k_{\text{pane}} = \frac{1.25t_{\text{pane}}^3}{\sum_i t_i^3} \leq 1$$

where

- k_{pane} = load-sharing factor of pane being checked
- t_{pane} = thickness of pane being checked (including laminated glass as per Clause 3.4.1 and glass thickness as per Clause 3.6 or Table 4.1)
- t_i = thickness of each pane of glass within the assembly (see Clause 3.6)
- i = total number of panes within the assembly

NOTE: For insulating glass units with two panes of equal thickness $k_{\text{pane}} = 0.625$.

Deflection of each panel is $=22.15\text{mm} \times 0.625=13.84\text{mm} < 20.83\text{mm}$ OK OK

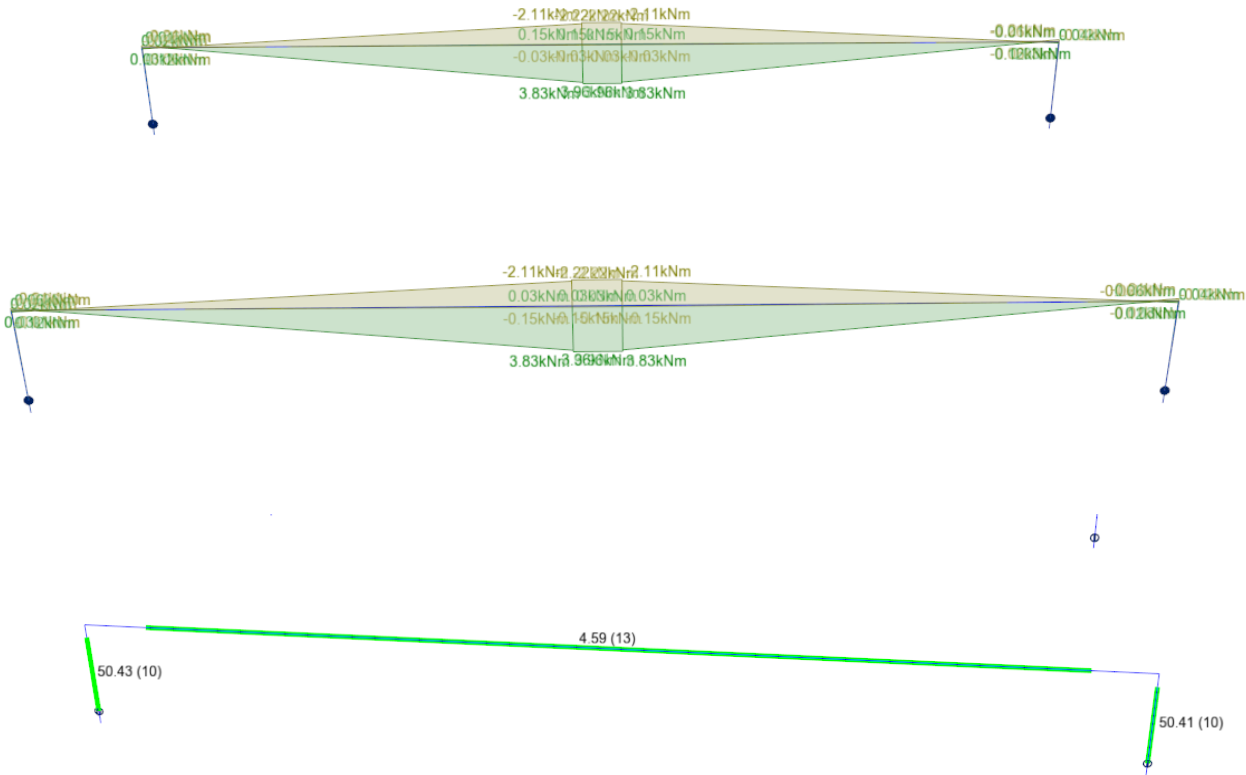
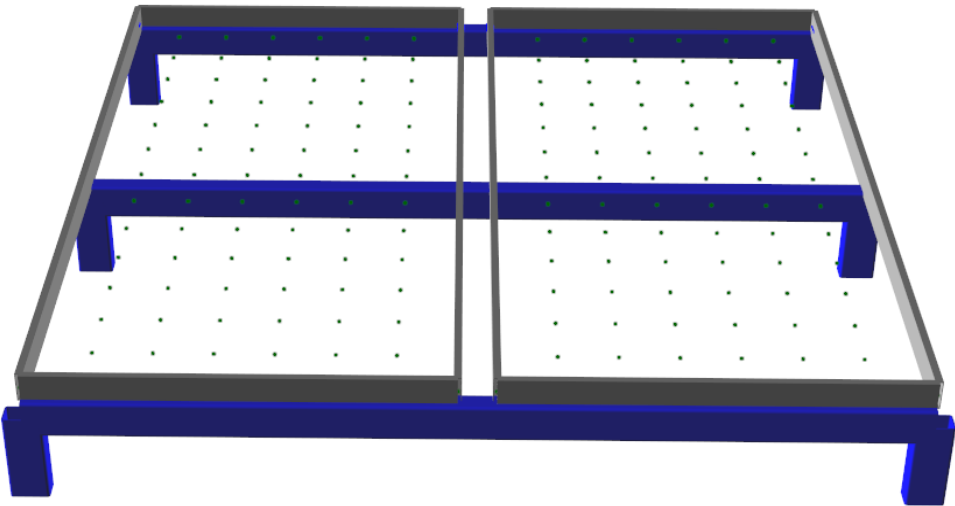
<div>  DETAILS CONSULTANT </div>	Project:	School hub Skylights (Junior)	Sheet:	16
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Aluminum frame Calculation

Max stresses in mid Aluminum frame is 0.69 KN.m Adopt 80x50x3 RHS Aluminum section OK

Check sheet No17-18

Steel frame Calculation



COMBINED SINGLE AXIS BENDING AND AXIAL- ALUMINUM BOX TUBE

Member 80x50x3 RHS

Basic Geometry and Design Action

$$\begin{aligned} M^* &= 0.6 \text{ kNm} & N^* &= 0 \text{ kN} \quad (\text{compression is positive sign}) \\ F_{L.B}^* &= 34.594 \text{ MPa} & F_{L.B}^* &= 0 \text{ MPa} \quad (\text{compression}) \end{aligned}$$

$$\begin{aligned} L_b &= 1380 \text{ mm} \quad (\text{unbraced length for bending}) \\ C_b &= 1.0 \quad (1.0 \text{ uniform moment, } 0.77 \text{ UDL to unbraced compression flange} \\ &\quad 1.28 \text{ UDL to compression flange between continuous restraints}) \end{aligned}$$

$$\begin{aligned} L_x &= 1380 \text{ mm} & L_y &= 1380 \text{ mm} \quad (\text{unsupported length for compression}) \\ k_x &= 1.00 & k_y &= 1.00 \quad (\text{effective length factor}) \end{aligned}$$

Trial Section and Alloy Properties

80	x	50	x	3.00	RHS
----	---	----	---	------	-----

$$\begin{aligned} A_g &= 744 \text{ mm}^2 & Z_c &= 16.19 \times 10^3 \text{ mm}^3 & E &= 70000 \text{ MPa} \\ I_x &= 0.65 \times 10^6 \text{ mm}^4 & r_x &= 30 \text{ mm} & F_{ty} &= 241 \text{ MPa} \quad (\text{refer AS 1664.1-1997 Table 3.3}) \\ I_y &= 0.308 \times 10^6 \text{ mm}^4 & r_y &= 20 \text{ mm} & F_{tu} &= 261 \text{ MPa} \quad (\text{Alloy and temper is 6061-T6}) \\ J &= 0.5 \times 10^6 \text{ mm}^4 & x_o &= 40 \text{ mm} & F_{cy} &= 241 \text{ MPa} \end{aligned}$$

Coefficients

$$\begin{aligned} \phi_y &= 0.95 & k_t &= 1.0 & B_c &= 271 \text{ MPa} \\ \phi_b &= 0.85 & k_c &= 1.0 & D_c &= 1.7 \text{ MPa} \\ \phi_u &= 0.85 & & & C_c &= 66 \end{aligned}$$

Bending Capacity

AS 1664.1:1997 Clause 3.4.15

$$\begin{aligned} S &= 111.63 (L_b \cdot Z_c) / (0.5 \sqrt{I_y J}) & \phi F_L &= 229 \text{ MPa (a) for } S < S_1 \\ S_1 &= 0.39 & & 206 \text{ MPa (b) for } S_1 \leq S \leq S_2 \\ S_2 &= 1696 & & 2055 \text{ MPa (c) for } S > S_2 \end{aligned}$$

$$\phi F_{L.B} = 206.15 \text{ MPa} \quad \text{OK}$$

Tension Capacity

AS 1664.1:1997 Clause 3.4.2

$$\phi F_{L.T} = 221.85 \text{ MPa} \quad \mathbf{N/A}$$

Compression Capacity

AS 1664.1:1997 Clause 3.4.8

$$\begin{array}{llll} \lambda_x = 0.8737 & D_c^* = 90 \text{ MPa} & \phi_{cc} = 0.76 & \phi F_L = 182.52 \text{ MPa (a) for } \lambda < S1^* \\ \lambda_y = 1.2667 & S1^* = 0.3326787 & & 119 \text{ MPa (b) for } S1^* \leq \lambda \leq S2^* \\ \lambda = 1.2667 & S2^* = 1.2306221 & & 114 \text{ MPa (c) for } \lambda > S2^* \end{array}$$

$$\phi F_{L.C} = 114 \text{ MPa} \quad \mathbf{OK}$$

Combined Tension and Bending

AS 1664.1:1997 Clause 4.1.2

$$\frac{0}{222} + \frac{34.59}{206.2} = 0.17 \quad \mathbf{N/A}$$

Combined Compression and Bending

AS 1664.1:1997 Clause 4.1.1

$$\begin{array}{ll} C_{mx} = 0.6 & \text{(simply supported, braced)} \\ F_{ex} = 114 \text{ MPa} & \end{array}$$

$$C_m = 0.6 - 0.4(M_1/M_2)$$

$$\frac{0}{114} + \frac{34.59}{206.2} = 0.17 \quad \mathbf{OK}$$

$$\frac{0}{114} + \frac{0.6}{206} \times \frac{34.594}{\left(1 - \frac{0}{114}\right)} = 0.10 \quad \mathbf{OK}$$